

Single and Multi-Channel

0906941

EGT, CHT, Carb Temp and OAT Instruments Operating and Installation Instructions

OII 0906941

9/6/94 *

You must read this manual before installing or operating the instrument. This manual contains warranty and other information that may affect your decision to install this product and/or the safety of your aircraft.



(This manual covers 25 instruments and remote switches)

(Supersedes OI 070781, II 070781, OI 040831 and II 070781-1)

(FAA Approved)

S/N: _____



Electronics International Inc. ®

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Important Notice

******* MUST READ *******

If you think it is not important to read this manual, you're wrong! This manual contains important installation information that may affect the safety of your aircraft, delay your installation or affect the operation of your instrument. You Must read this manual prior to installing your instrument. Any deviation from these installation instructions is the sole responsibility of the installer/pilot and may render the STC invalid.

Read the Warranty / Agreement. There is information in the Warranty / Agreement that may alter your decision to install this product. **If you do not accept the terms of the Warranty / Agreement, do not install this product.** This product may be returned for a refund. Contact Electronics International inc. for details.

Check that the instrument make and model marked on the side of the instrument and on the invoice are correct before starting the installation.

It is possible for any instrument to fail thereby displaying inaccurate high, low or jumpy readings. Therefore, you must be able to recognize an instrument failure and you must be proficient in operating your aircraft safely in spite of an instrument failure. If you do not have this knowledge, contact the FAA or a local flight instructor for training.

The ability for this product to detect a problem is directly related to the pilot's ability to program proper limits and the pilot's interpretation and observation skills.

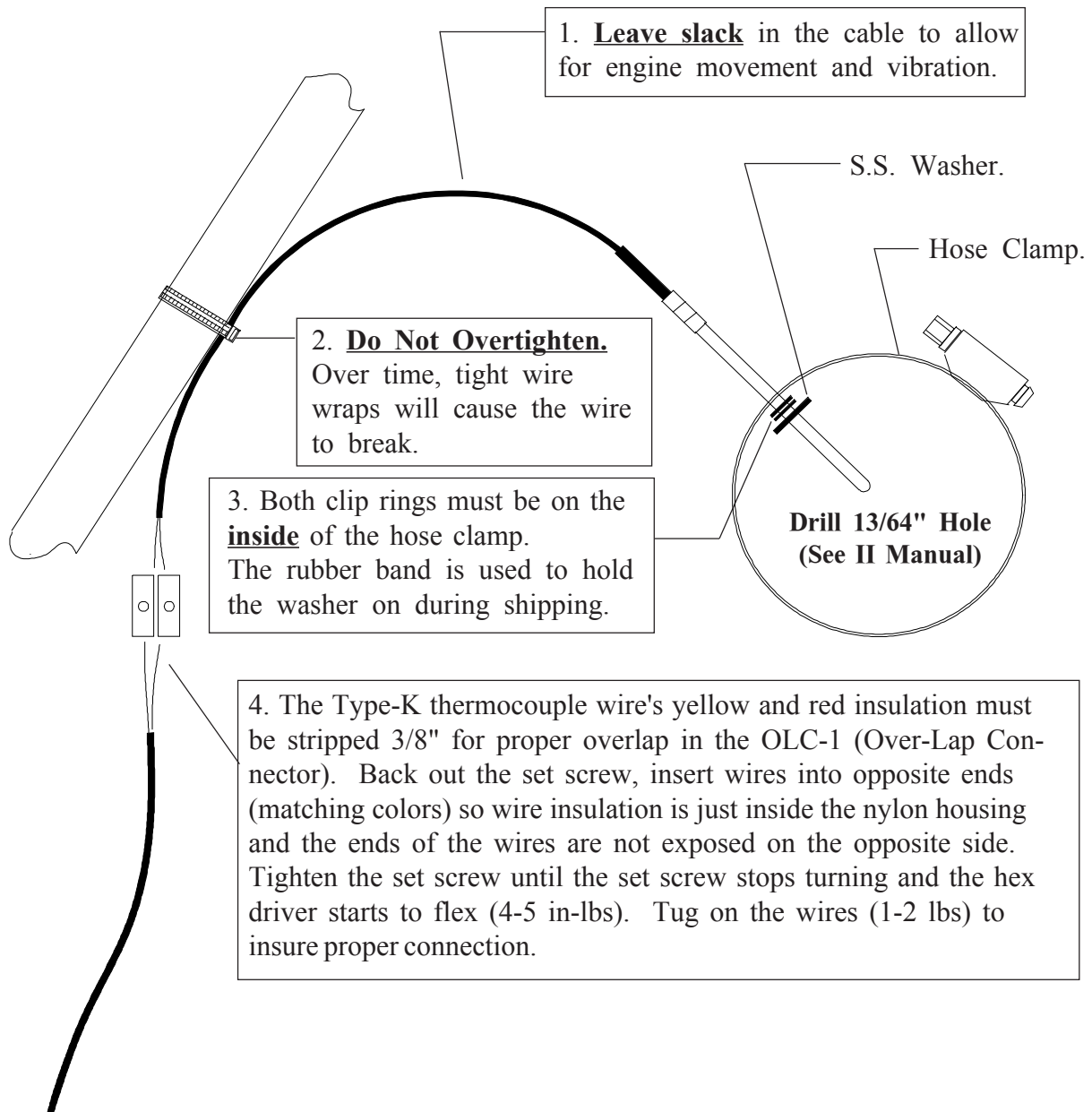
The pilot **must** understand the operation of this product before flying the aircraft. Do not allow anyone to operate the aircraft that does not know the operation of this product. **Keep the Operating Manual in the aircraft at all times.**

P-110 Standard EGT Probe

1218121

Important Installation Information

All steps must be read before installing a probe.

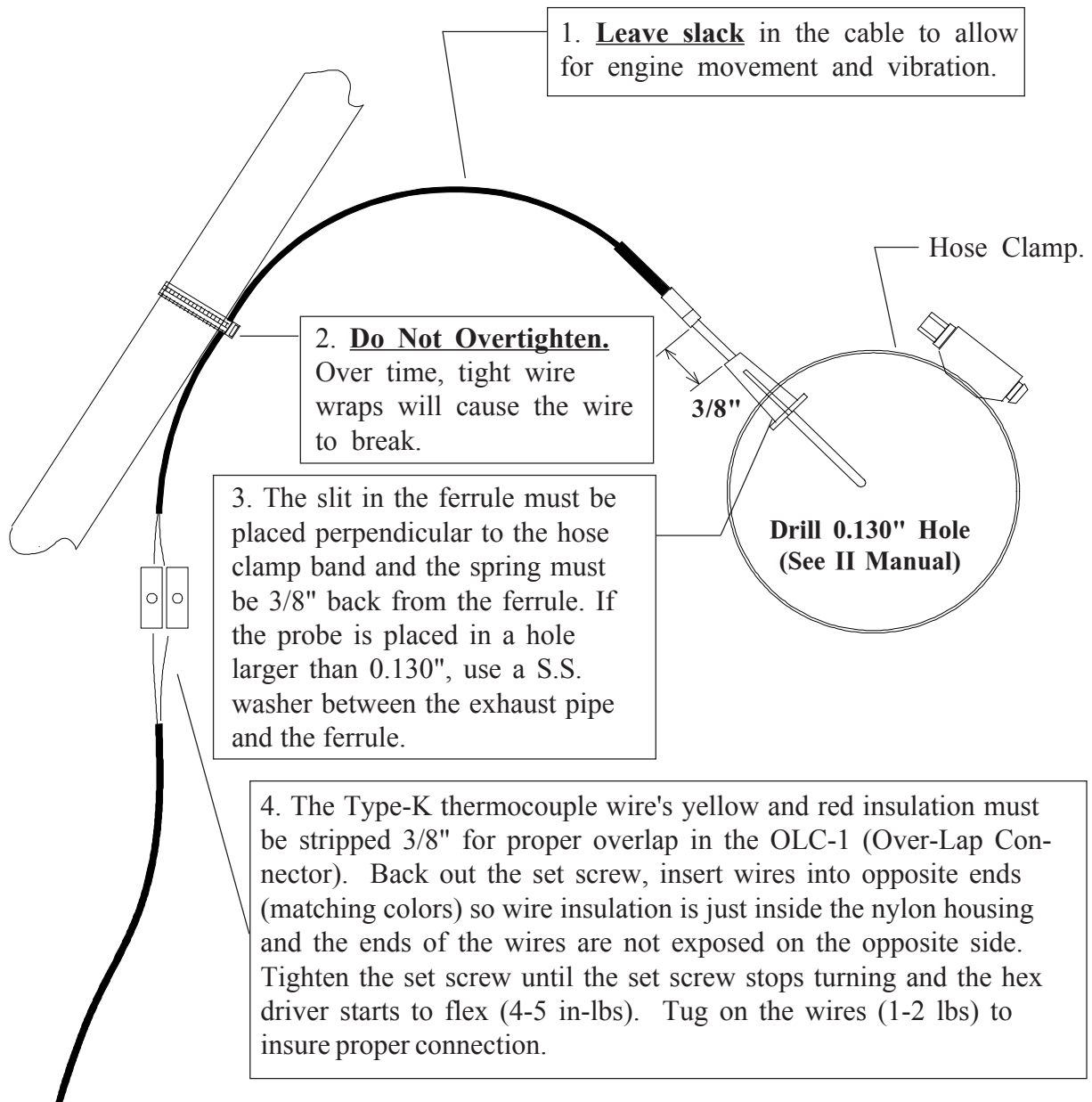


P-110 Fast Response EGT Probe

1218122

👉 Important Installation Information 👈

All steps must be read before installing a probe.



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Warranty / Agreement

Electronics International Inc. warrants this instrument and system components to be free from defects in materials and workmanship for a period of one year from the user invoice date. Electronics International Inc. will repair or replace any item under the terms of this Warranty provided the item is returned to the factory prepaid.

1. This Warranty shall not apply to any product that has been repaired or altered by any person other than Electronics International Inc., or that has been subjected to misuse, accident, incorrect wiring, negligence, improper or unprofessional assembly or improper installation by any person. **This warranty does not cover any reimbursement for any person's time for installation, removal, assembly or repair.** Electronics International retains the right to determine the reason or cause for warranty repair.

2. This warranty does not extend to any machine, vehicle, boat, aircraft or any other device to which the Electronics International Inc. product may be connected, attached, interconnected or used in conjunction with in any way.

3. The obligation assumed by Electronics International Inc. under this warranty is limited to repair, replacement or refund of the product, at the sole discretion of Electronics International Inc.

4. Electronics International Inc. is not liable for expenses incurred by the customer or installer due to factory updates, modifications, improvements, upgrades, changes, or any other alterations to the product that may affect the form, fit, function or operation of the product.

5. Personal injury or property damage do to misinterpretation or lack of understanding this product is solely the pilots responsibility. The pilot **must** understand the operation of this product before flying the aircraft. Do not allow anyone to operate the aircraft that does not know the operation of this product. Keep the Operating Manual in the aircraft at all times.

6. E. I. Inc. is not responsible for shipping charges or damages incurred under this Warranty.

7. No representative is authorized to assume any other liability for Electronics International Inc. in connection with the sale of Electronics International Inc. products.

8. **If you do not agree to and accept the terms of this warranty, you may return the product for a refund.**

This Warranty is made only to the original user. **THIS WARRANTY IS IN LIEU OF ALL OTHER WARRANTIES OR OBLIGATIONS: EXPRESS OR IMPLIED. MANUFACTURER EXPRESSLY DISCLAIMS ALL IMPLIED WARRANTIES OF MERCHANTABILITY OR FITNESS FOR A PARTICULAR PURPOSE. PURCHASER AGREES THAT IN NO EVENT SHALL MANUFACTURER BE LIABLE FOR SPECIAL, INCIDENTAL OR CONSEQUENTIAL DAMAGES, INCLUDING LOST PROFITS OR LOSS OF USE OR OTHER ECONOMIC LOSS. EXCEPT AS EXPRESSLY PROVIDED HEREIN, MANUFACTURER DISCLAIMS ALL OTHER LIABILITY TO PURCHASER OR ANY OTHER PERSON IN CONNECTION WITH THE USE OR PERFORMANCE OF MANUFACTURER'S PRODUCTS, INCLUDING SPECIFICALLY LIABILITY IN TORT.**

OPERATING INSTRUCTIONS

Features

1. Instruments

The Electronics International line of single and multi-channel instruments offer the following features:

A. Digital Display - The digital display allows you to read absolute temperatures at a glance. It does not require interpretation of dials or tic marks. In a short period of time you will become familiar with the normal operating temperatures of your engine. Abnormal temperatures will be easy to spot. The digital display is easily viewable in direct sunlight. If the instrument backlight has been permanently powered up (as recommended), the digital display will be easier to see during low ambient light conditions and at night.

B. 1 Degree Resolution - The digital display resolves temperatures to 1 degree. This allows you to interpret trends quickly. This can be very helpful in diagnosing problems and leaning your engine. Also, any unit may be ordered to display in degrees F or degrees C.

C. 1/2% Accuracy - Electronics International instruments are not affected by shake, shock, vibration, tilt, stick-slip, bearing wear, spring wear, lead resistance, probe resistance, magnetic fields or the many other factors that plague analog instrument accuracy. All E.I. instruments are temperature compensated to read cabin temperature when a probe is disconnected. E.I. instruments should never need recalibration.



Ice Zone Warning Light. Only available on the Carb. Temp/OAT instruments.

Digital Display.

Selector Switch.

D. Flexibility - Electronics International instruments are compatible with any type K ungrounded probe. This means any instrument, regardless of what is printed on the front panel (EGT, CHT, OAT, etc.), will work with any of our probes (i.e., an OAT channel can read EGT or CHT probes accurately). Also, lead resistance does not affect the accuracy of these units. You may use any length extension cable between the unit and the probe without affecting the accuracy of the instrument.

E. Upgradable - Any single channel EGT and/or CHT unit may be upgraded to a full multi-channel analyzer by simply adding a remote switch to the system. The instruments, remote switches, extension cables and probes were designed in a modular fashion with slip-on connectors. This means a remote switch may be added to your existing system by simply mounting it into your instrument panel, installing the additional wires and probes and plugging it in. You do not have to buy a new system to upgrade to a full analyzer.

2. Remote Switches



Instrument displaying
EGT for channel #2.

A remote switch may be connected to any channel on any Electronics International Instrument. This gives the instrument multi-channel capability. There are two types of remote switches available, single deck and double deck. If a single deck four channel remote switch (RS-4-1) is connected to the EGT channel on an EC-1, the instrument would be capable of measuring four EGTs and one CHT. The remote switch will select which EGT channel would be displayed on the EC-1 instrument when the instrument's selector switch is placed in the EGT position. If a double deck four channel remote switch (RS-4-2) is connected to the EGT and CHT channels on an EC-1, the instrument will be capable of measuring four EGTs and four CHTs. The EC-1 will select whether EGT or CHT will be displayed. The Remote Switch will select which channel will be displayed.

EGTs

1. Leaning

You will want to lean your engine in cruise. A rich running engine wastes fuel needlessly and tends to run rough. This creates vibration, which causes deterioration of engine accessories and engine mounts. Also, proper leaning at cruise and during descent means less spark plug fouling, longer life for the plugs, reduced maintenance costs and considerable fuel savings. Furthermore, good leaning techniques result in cleaner combustion chambers with fewer lead salt deposits on the pistons and exhaust valves. Under certain conditions, these deposits invite preignition and higher maintenance costs. Proper leaning at cruise during cool or cold weather aids in raising engine and oil temperatures to desirable minimums in order to evaporate the water and acids out of the oil. Water and acids attack the insides of an engine, causing rust and corrosion.

To properly lean your engine using a multi-channel analyzer perform the following steps:

A. Rough Leaning: Select the hottest EGT cylinder. Adjust the mixture control from the full rich position to a leaner setting that results in a slight drop in engine RPM or to a setting near peak EGT, as dictated by experience. The mixture control should be left at this setting until the EGT's stabilize. It will take about 20 seconds for the temperatures to stabilize within 1°F. This lag is due to the combustion walls and piston domes increasing in temperature, which affect the combustion and exhaust gas temperatures. **To correctly lean an engine you must wait for the engine to thermally stabilize.** Less sensitive gauges will not pick up these subtle changes, which are important in leaning and diagnosing problems.

B. Precision Leaning: Again select the hottest EGT cylinder. This cylinder may be different than the one you started with. This is the cylinder on which you should perform your precision leaning. Again, start leaning, making only very small adjustments and waiting 3 to 5 seconds between adjustments. As you approach peak, the exhaust gas temperature will rise much slower until it starts to decrease. When this happens you have reached peak EGT. The

1°F resolution of the digital display will be invaluable in helping you precisely detect peak EGT.

C. Finding The Cylinder That Peaks First: For most engines Step B (Precision Leaning) will result in a properly leaned engine. If you find this to be the case with your engine, this step will not be necessary. But if you want to verify that you have leaned to the cylinder that peaked first and your engine is operating properly, perform the following with the cylinder found in step B at peak EGT. Slightly enrich the mixture and quickly step through each cylinder. Any cylinder that shows a rising temperature is a leaner cylinder. Check that this cylinder does not rise more than 15°F before it starts decreasing in temperature. If a cylinder rises more than 15°F it may have a problem.

When installing a single channel EGT instrument in an aircraft there is no guarantee that the probe is installed on the leanest cylinder. Every engine operates a little differently. For the same make and model of engine installed in the same type of aircraft there can be differences between the leanest cylinders. Furthermore, there can be a difference between operating temperatures and the temperature spread between cylinders. Every engine has its own unique operating temperatures. To properly lean your engine using a single channel EGT unit perform the following steps:

A. Rough Leaning: Adjust the mixture control from the full rich position to a leaner setting that results in a slight drop in engine RPM or to a setting near peak EGT, as dictated by experience. The mixture control should be left at this setting until the EGT's stabilize. It will take about 20 seconds for the temperatures to stabilize within 1°F. This lag is due to the combustion walls and piston domes increasing in temperature and, therefore, affecting the combustion and exhaust gas temperatures. **To correctly lean an engine you must wait for the engine to thermally stabilize.** Less sensitive gauges will not pick up these subtle changes, which are important in leaning and diagnosing problems.

B. Precision Leaning: Again, start leaning, making only very small adjustments and waiting 3 to 5 seconds between adjustments. As you approach peak the exhaust gas temperature will rise much slower until it starts to decrease. When this happens you have reached peak EGT. The 1°F resolution of the digital display will be invaluable in helping you precisely detect peak EGT. You will then need to enrich the mixture for an EGT reading 30°F lower than peak to insure there is no cylinder operating on the lean side of peak EGT.

If your engine runs rough before peak EGT is reached, note the temperature reading on the EGT instrument. When an engine starts to run rough (not when it loses power, but actually runs rough) the leanest cylinder has gone past peak EGT by 30 to 50 degrees F. The leanest cylinder is lean misfiring causing the engine to run rough. From this point enrich the mixture to obtain a 50 degrees F lower EGT from the noted temperature. This will set the leanest cylinder slightly on the rich side of peak EGT. The rest of the cylinders will be running richer than the leanest by an amount dictated by the temperature spread for your engine. With this method you can reasonably lean an engine even when the probe has not been mounted on the leanest cylinder.

Electronics International's unique stable display allows you to precisely lean to peak EGT or to a specific temperature below peak for most engines. Peak EGT with a float-type carbureted engine is frequently a vague point because of the fuel/air distribution issues in these lower horsepower

engines. As a result, these engines tend to operate smoother at 25°F on the rich side of peak EGT. Fuel-injected engines will provide a more precise peak. Most engines normally operate within an EGT range of 1300°F to 1600°F at cruise power.

Some engine manufacturers allow leaning to peak EGT at 75% power and below on their direct drive normally aspirated engines. **For your engine, check the engine manufacturer's recommended procedures. It is not recommended to lean to peak EGT above 75% power settings.** The richer mixture is needed to cool the combustion temperatures and keep the anti-knock capability of the fuel high enough to prevent detonation from occurring at the higher power settings.

2. EGT Diagnostics

Since the EGT is directly related to the combustion temperature, it is an indication of the engine's ability to produce power. If the engine is not producing the correct amount of power, the EGT instrument can be a very valuable troubleshooting tool as well an early warning system before engine failure occurs. With 1°F resolution, our digital EGT instruments will react to the slightest changes in the combustion process. To detect a problem, become familiar with your engine's normal EGT readings during run-up, climb, cruise and descent. Any difference from the norm can be a sign of trouble.

During normal operation the EGT will stabilize to 1°F for a given throttle and mixture setting. If it does not stabilize, this can also be the first sign of trouble. With rate and trend information being displayed instantaneously and with temperatures being read to 1°F, few problems can escape the pilot flying one of Electronics International's analyzer systems. The following is a list of EGT/CHT symptoms and possible problems:

| Symptom | Possible Problem |
|--|---|
| One EGT reads abnormally high. The corresponding CHT reads lower than normal. | — Burned valve or broken ring, defective plug, plug wire or mag. |
| One EGT reads abnormally high. The corresponding CHT reads higher than normal. | — Plugged injector, intake leak. |
| One EGT reads abnormally low. | — Over-sized injector, restricted exhaust, broken or leaky exhaust header. |
| High CHTs and/or high EGTs on all cylinders. | — Excessive leaning with power settings over 75%. Detonation due to bad fuel. Closed or restricted cowl flaps. Missing or loose baffling. |
| High EGTs and/or low CHTs on all channels. | — Timing problem or defective mag. |
| Jumpy readings on one channel. | — This is not an engine problem. Check all connections and the probe for proper operation. See Troubleshooting Section of this manual. |

It is not necessary to continually monitor the EGTs in order to detect a problem. Most problems worsen over a period of time and can be easily detected before they become a safety hazard by thoroughly checking the EGT readings at run-up and once or twice during a flight.

CHTs

1. CHT Operation

The Cylinder Head Temperature (CHT) instrument helps the pilot protect his engine against the threat of excessive heat. Most general aviation aircraft monitor the hottest CHT, as determined by extensive flight tests done by the airframe manufacture. Minimum in-flight CHT should be 150°F, and maximum in most direct drive normally aspirated Avco Lycoming engines is 500°F. Some of the higher powered, more complex engines have a limit of 475°F. Although these are minimum and maximum limits, the pilot should operate the engine at more reasonable temperatures in order to achieve the expected overhaul life of the powerplant. It would be normal during all-year operations in climb and cruise to see cylinder head temperatures in the range of 350°F to 435°F.

Sudden cooling of the CHT (known as shock cooling) is a problem that is common with aircraft engines. This is caused by fast descents with little or no power and rich mixtures. This may result in bent pushrods due to exhaust valves sticking, burned valves, spark plug fouling, broken piston rings, cracked cylinders at the spark plug and valve ports and warped exhaust valves. To avoid these problems, do not allow the CHT to cool more rapidly than 1°F every 3 seconds during in-flight operation. This can be easily detected with our 1°F digital display.

During climbs, the cylinder head temperatures will rise rapidly until the heat absorbed by the combustion walls is dissipated out the engine's cooling fins. At this point, the CHT will stabilize. Any change in throttle, mixture, cowl or airspeed will affect the CHT and the rate at which it will change. Since rate and trend information can be easily interpreted from our digital display, changing any one of these parameters to stabilize, slow or reduce the CHT is possible with almost immediate results. Our digital instrument takes the guesswork out of controlling your CHT.

2. CHT Diagnostics

The source of heat in an engine is from the combustion of the fuel/air mixture producing temperatures of approximately 4000°F. Some of this heat energy goes into heating the cylinder heads through radiation and conduction. This heat is sinked away from the engine by the air flow over the cylinder heads. When the heat being generated in the cylinder heads equalizes with the heat being sinked away, the cylinder head temperature will stabilize. If a problem arises in the combustion chamber or in the ability of the cooling system to sink away heat, the CHTs will be affected. To detect a problem, become familiar with your engine's CHT operating temperatures during run-up, climb, cruise and descent. Any differences from normal can be a sign of trouble.

Continuous change in the CHT can also be a sign of trouble. Because of the large thermal mass of the engine, the CHTs change slowly after the initial climb. Any continuous change in one or all of the CHTs after this initial climb can be the sign of trouble. The rate and trend of this change can easily be detected with Electronics International's 1°F resolution digital display. This information allows the pilot to make changes in flight attitude or engine operation and see the effects almost instantaneously.

Carburetor Temperature

Venturi affect and atomization of fuel can cause temperatures in the carburetor to drop 30°F or more. When the atmospheric conditions are right for the aircraft's current flight altitude (moderate to high humidity), the moisture in the carburetor venturi can freeze quickly. Within minutes ice can choke off the venturi and the engine will stop with little warning.

When Carb. Temp. is selected on the Electronics International Carb. Temp. instrument, the carburetor temperature is continuously monitored and the "Ice Zone" warning light over the display is activated for that channel. The "Ice Zone" warning light is only active for the channel selected. At 39°F (before ice can form in the venturi of the carburetor) the "Ice Zone" warning light will light up. When this happens, apply carburetor heat, making small adjustments to bring the carburetor temperature above 39°F, thereby avoiding any possible carburetor icing condition. An additional benefit of running carburetor temperatures 9°F above freezing is improved atomization of the fuel which results in fewer lead deposits, cleaner plugs and better economy. If the carburetor temperature is below 10°F the "Ice Zone" warning light will go off. Below 10°F there is not enough moisture in the air to form ice in the carburetor.

The "Ice Zone" warning light has the advantage of catching your attention without having to continuously monitor the unit. At night this light may be too bright. An LED Intensity Control Line is provided which may be connected to the aircraft panel rheostat. When the instrument panel lights are turned up the "Ice Zone" warning light will dim.

Monitoring carburetor temperature to 1°F can also help with hard to start engines. If the engine becomes flooded and fuel starts to drip from the carburetor, the unit will display a drop in carburetor temperature as the fuel starts to evaporate. If the engine backfires and a fire starts in the venturi, the unit will display a rapid rise in the carburetor temperature. The carburetor probe is rated for 700°F, so probe damage is not likely.

Outside Air Temperature

The Electronics International OAT instrument has three features that make it a valuable tool when measuring outside air temperatures. The first of these features is its superior accuracy and linearity over conventional gauges. Outside air temperatures have a big affect on your aircraft's ability to lift

and on engine horsepower. Accurate OAT readings are essential if you are looking for maximum performance from your aircraft.

The second valuable feature is the instrument's ability to detect small temperature changes (1°F). This gives the pilot rate and trend information (in what direction and how fast the temperatures are changing) at a glance. This is valuable for detecting changing atmospheric conditions and avoiding thunderstorms and icing conditions. It can also help to find cooler flying conditions in warm weather.

The third feature is the instrument's Ice Zone Warning Light. This light will come on when the OAT drops to 39°F and stays above 10°F. This feature can be very useful to a pilot by warning him of the possibility of structural ice if weather conditions are right.

The Electronics International OAT instrument resolves outside air temperatures to 1°F and is very sensitive to air temperatures changes. For this reason, when the OAT probe is in still air and near a heat source, such as hot asphalt, a hangar heater, etc., the unit will read the actual temperature to which the probe is subjected. When the engine starts and there is a flow to air over the probe, the unit will read the air temperature accurately and display changes quickly.

INSTALLATION INSTRUCTIONS

1. Important Information and Initial Check Out

A. The installer and aircraft owner must read the Warranty before starting the installation. There is information in the Warranty that may alter your decision to install this instrument. **If you do not accept the terms of the Warranty, do not install this instrument.**

B. If you are not an FAA Certified Aircraft Mechanic familiar with the issues of installing aircraft EGT, CHT, Carb Temp and/or OAT instruments, **Do Not attempt to install this instrument.** The installer should use current aircraft standards and practices to install this instrument (refer to AC 43.13).

C. Check that any necessary FAA Approvals (STC's, etc.) are available for your aircraft before starting the installation. STC's are located at the back of this manual.

D. Read the entire Installation Instructions and resolve any issues you may have before starting the installation. This may eliminate any delays once the installation is started.

E. THIS INSTALLATION MAY REQUIRE SOME PARTS UNIQUE TO YOUR AIRCRAFT THAT ARE NOT SUPPLIED IN THE KIT. Acquire all the parts necessary to install this instrument before starting the installation.

F. Check that the instrument make and model are correct before starting the installation.

G. Before starting the installation make sure the unit will fit in the location you intend to install it without obstructing the operation of any controls.

H. If this instrument is to replace an existing unit in the aircraft, it is the installer's responsibility to move or replace any existing instruments or components in accordance with FAA approved methods and procedures. The following Installation Instructions do not cover moving or the removal of any existing instruments or components.

2. CHT Probe Installation

A single CHT probe should be placed on the hottest cylinder. In a 6-cylinder engine this would be one of the center cylinders. On a 4-cylinder engine this would be one of the back cylinders.

If a second CHT probe is to be installed it should be placed on one of the front unobstructed cylinders. This will allow the unit to detect shock-cooling.

Most engines have a port just below the lower spark plug for the CHT probe. **If your engine has a primary CHT probe in one of the cylinders, do not remove it.** Select another cylinder for your probe. If you are putting a CHT probe on every cylinder use our P-102 Gasket CHT Probe for secondary readings on your primary cylinder.

3. EGT Probe Installation

A single EGT probe should be installed in the exhaust stack of the leanest cylinder. Each engine has its own characteristics and the leanest cylinder can be different from aircraft to aircraft. As a general rule, the leanest cylinder is one of the back cylinders on a carbureted engine and one of the center cylinders on a fuel-injected engine.

Look at each exhaust stack and determine the best location at which all of the EGT probes can be mounted at the same distance down from the exhaust ports. The ideal location is 1 1/2", but ease of installation should prevail. Drill a 13/64" diameter hole in each exhaust stack. Insert the probe and tighten the hose clamp. As the hose clamp is heated and cooled, it will become loose as it conforms to the exhaust stack. After the first 10 hours of operation, each hose clamp should be retightened.

IMPORTANT NOTE: For Cessna 210's or any aircraft using a slip joint in the exhaust system, install the EGT probes ABOVE OR BELOW THE SLIP JOINT. Installing a EGT probe in the slip joint can damage the probe.

4. Carb Temp Probe Installation

Remove the threaded plug located in the carburetor housing just below the throttle valve. Install the Carburetor Temperature Probe in this hole using a lock washer. Care should be taken not to over-tighten the probe and strip the threads in the carburetor housing.

If your carburetor does not have a factory tapped hole, remove the carburetor from the engine. Drill out the lead plug located just below the throttle valve with a 7/32 drill and tap this hole with a 1/4 x 28 tap. Remove all burrs and metal shavings from the interior and exterior of the carburetor. Install the carburetor temperature probe as outlined above. Reinstall the carburetor on the engine.

5. OAT Probe Installation

Mount the OAT Probe in an appropriate location on the aircraft, using the hardware supplied. The OAT Probe is sensitive to air temperature changes. For this reason, do not mount the OAT probe in the path of the cowl or engine exiting air (i.e., on the belly of the aircraft). Also, the probe should not be mounted within 8 inches of an exhaust pipe. The radiant energy from the exhaust pipe can cause the probe to read slightly high. Other than these considerations the OAT Probe may be mounted in an air intake vent, on the side of the cowl or anywhere else on the aircraft.

6. Route the Extension Cables

Mark both ends of each cable with the appropriate cylinder and/or probe type. Strip wires and connect probe to extension cable, matching wire colors, using the supplied OLC-1 (Over-Lap Connectors). When tie wrapping these cables down, be sure there is no strain or pulling on the cable against the probe housing. Each probe should have 3 to 4 inches of slack to allow the engine to move in its mount without breaking any wires. Also, there should be a tie wrap near the probe connectors. Dress each cable up to the instrument or remote switch keeping them away from any hot areas (exhaust stacks, cylinder heads, etc.).

If you must shorten the cable, pull any excess cable length through the fire wall and cut it off at this time. However, it is recommended you leave some extra wire length under the instrument panel for later modifications. Varying cable lengths will **not** affect the accuracy of this instrument so cables of any length may be ordered from the factory. The Extension Cables and probe wires are made of type K thermocouple wire that **must not be substituted or extended with regular copper wire**. Also, it is important these wires not be kinked (i.e., **do not bend the wires on a radius less than 1 inch**).

Connect the Extension Cables using the OLC-1 connectors as described below:

- A. Strip back the insulation on the two mating wires 3/8". Be careful not to nick the wires.
- B. Place the hex driver (0.050") in the OLC-1 set screw and back out the set screw to the last thread.
- C. Inset the two wires into the ends of the connector so the wire insulation is just inside the nylon housing and the ends of the wires are not exposed on the opposite side.
- D. Tighten the set screw until the set screw stops turning and the hex driver starts to flex (4 to 5 in.-lbs.)
- E. Tug on the wires (1 to 2 lbs.) to insure proper connection.

Tie off any excess cable under the instrument panel. Be sure these cables do not obstruct the freedom of travel of any controls. Cable length does not affect the accuracy of our units, so cables of any length may be ordered from the factory.

7. Hook-Up Connecting Wires

Connect the red wire in the harness to the 12 or 24 volt bus via a 1 amp fuse or circuit breaker (see the Wiring Diagram at the back of this manual). Connect the black wire in the harness to ground. Connect the white/violet to the 12V bus (leave it open for a 24V system). Connect the white/gray wire to the 24V bus (**connect to ground for a 12V system**). If this unit has an "Ice Zone" warning light, connect the white/orange wire to the Panel Light Rheostat to dim the "Ice Zone" warning light at night.

Tie off any excess cable under the instrument panel. Be sure these cables do not obstruct the freedom of travel of any controls.

8. Connect and Install the Instrument and/or Remote Switch

Connect the extension cables and all other wires to the instrument and/or remote switch as shown in the appropriate wiring instructions at the back of this manual. **Be sure the OLC-1 connectors are installed properly.** Install the unit from behind the instrument panel using 6 x 32 screws. These screws must not be longer than 0.4 inches. Install the remote switch through a 1" square hole cut into the aircraft instrument panel. Tie wrap cables as needed. Be sure these cables do not obstruct the freedom of travel of any controls.

9. System Ground Test

A. Turn the master switch on and look for a near ambient temperature reading on each channel. If the instrument does not power-up (display a reading), check the power and ground leads (red and black leads) for an open, loose or poor connection.

If you suspect any channel is not receiving a signal remove the probe from the engine (leaving it connected to the Extension Cable) and apply a temperature to it. Look for an increase in reading on the display for that channel. Check the other channels for an increase in reading. You may have connected the probe to the wrong Extension Cable. If the reading is decreasing, you may have reversed the connectors on the Extension Cable leads (the yellow wire on the probe must connect to the yellow wire on the Extension Cable).

B. Start the engine and check each channel for a proper reading. On the ground EGTs will read around 900°F and CHTs will read around 200°F. If you suspect any channel is not receiving a signal properly, see the "Troubleshooting" section of this manual.

TROUBLESHOOTING SUGGESTIONS

Because high reliability is designed into Electronics International's equipment, there is no reason to put up with poor operation. We have few problems with our probes, cables and units and installation is simple. Usually fixing a problem is just a matter of inspecting the installation at a few key points.

Strategy

If you have more than one problem, **FIX ONE PROBLEM AT A TIME**. Trying to fix all of them at once can be confusing and misleading. In many cases fixing one problem first will lead you to the solution for fixing all of the problems. Therefore, take one problem on one channel and proceed with the following:

1. Instrument Check Out

If there is an identical symptom on each channel, then the instrument may have a problem. But if even one channel of the instrument is operating properly, the instrument probably does not have a problem. A good method to test the instrument is to remove all the Extension Cables. Then look for a reading on all channels to be near cabin temperature. The only inputs a unit requires to operate properly and measure cabin temperature is power (red lead) and ground (black lead). Check the power and ground leads for proper connection (pull on the wire at each connector).

NOTE: Few problems turn out to be the instrument.

2. Probe Check Out

There are two good methods of testing a probe. Perform one or both of the following:

A. A probe can be tested with an ohmmeter. Disconnect the probe from the Extension Cable. When testing the resistance between the connectors, the probe should measure a "short" (less than 5 ohms). When measuring from one lead (either lead) of the probe to the probe sheath (metal tip), there should be an "open" (10kohm or greater).

B. Another method of checking a probe is to plug the suspected bad probe into a channel that is working properly. If the problem follows the probe, you have a defective probe.

3. Extension Cable Check Out

With the Extension Cable connected to the unit, remove the probe from the suspected bad Extension Cable. Set the unit to the proper channel and look for a near cabin temperature reading on the display. A very high or low reading indicates a short to ground in the cable. Next, connect an

ohmmeter, set to 10K range, to the open probe ends of the suspected bad Extension Cable. Set the unit to the proper channel and look for a very high (+ or -) reading on the digital display. A near cabin temperature reading or no change in reading indicates an open in the cable or its connectors. Also, look for a reading on the ohmmeter around 12K ohms. Most problems of this kind are usually one of the following:

- A. Improper OLC-1 Connections:** Pull on wires installed in the Over-Lap Connector to check the connection. You may have insulation in the overlap area. Remove the wires from the OLC-1 and inspect.
- B. Broken Wire:** A wire can be broken from a too-tight tie-wrap or by repeatedly flexing the wire. Inspect the wires for a break. Note: A wire can be broken while the insulation is still intact.
- C. Cable Chafed to Ground:** If a cable is routed around a metal object, it will over time chafe the wire and short to the object. Inspect the wires for chafing.

SPECIFICATIONS and OPERATING FEATURES

S0908941

9/8/94

Models: (Temperature Monitoring Instruments) E-1, C-1, A-1, CA-1, EAC-1, EAE-1, EC-1, EC-2, E-2, C-2, E-4, C-4, E-6 and C-6.

Weight: E-1, C-1, A-1, CA-1, EAC-1, EAE-1, EC-1, EC-2, E-2 and C-2 - 7 oz. (unit only)
E-4 and C-4 - 7.6 oz. (unit only)
E-6 and C-6 - 8.3 oz. (unit only)
6' T.C. Cable - 2 oz.
EGT or CHT Probe - 1.6 oz.

Environmental: Meets TSO C43a

Power Requirements: 7.5 to 30 Volts, 1/10 Amp.

Display: LCD's (viewable in direct sunlight), with 12 and 24 volt backlight control wires.

Ice Zone Warning Light: This light comes on when the displayed temperature is between 39°F and 10°F. An Intensity Control Line is provided to dim this light for night operation.

Display Temperature Range: 1999°F to -1999°F

Accuracy: 1/2% in accordance with TSO C43a.

Resolution: 1°F (with enhanced stability and response circuits).

Probes: Type K, Ungrounded (for improved accuracy, stability and reliability).

Extension Cables: Type K, any length or size (you may use your existing type K cables).

Models: (Remote Switches) RS-4-1, RS-4-2, RS-6-1, RS-6-2, RS-8-1 and RS-12-1

Weight: RS-4-1, RS-4-2, RS-6-1 and RS-8-1 - 3.2 oz. (unit only)
RS-6-2 and RS-12-1 - 4.9 oz. (unit only)

Environmental: Meets TSO C43a

Contacts: Gold

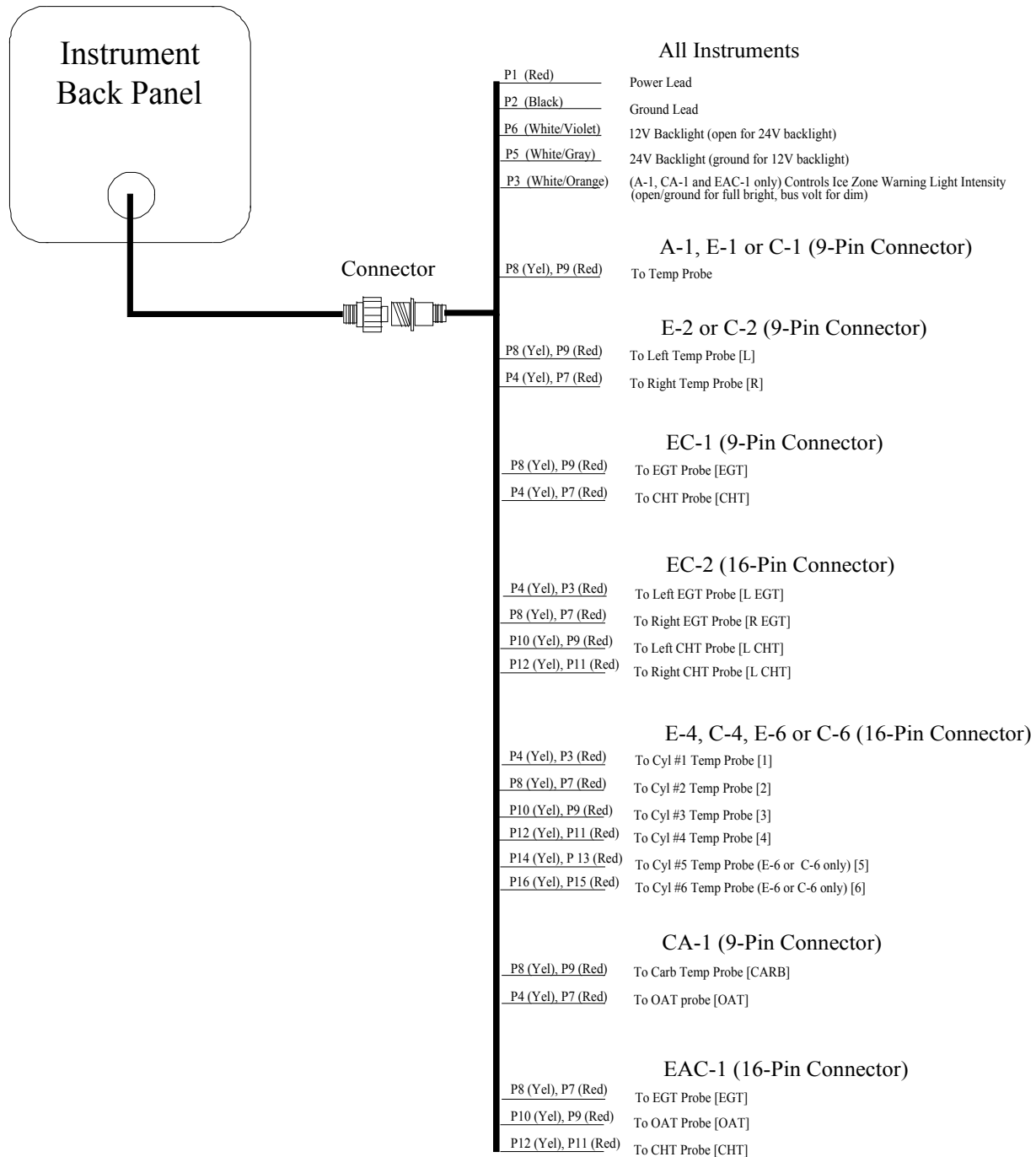
Contact Resistance: <50 milliohms

Angle of Throw: 30 degrees

Rotational Torque: 3.5 to 7.5 ounce-inches

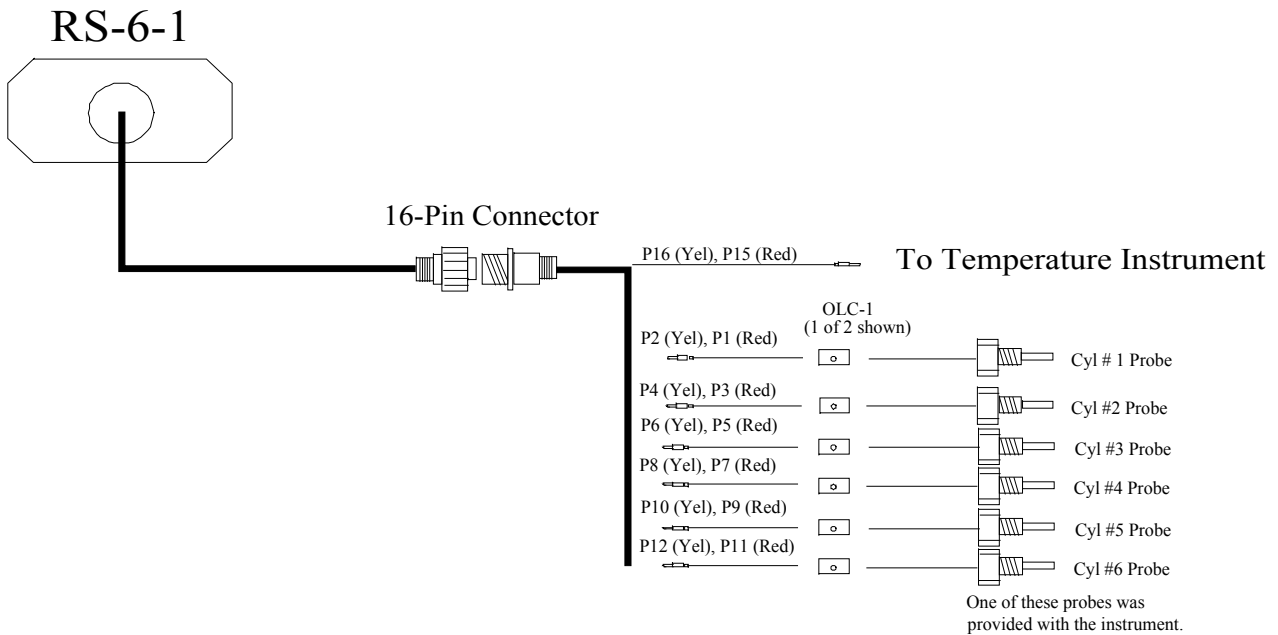
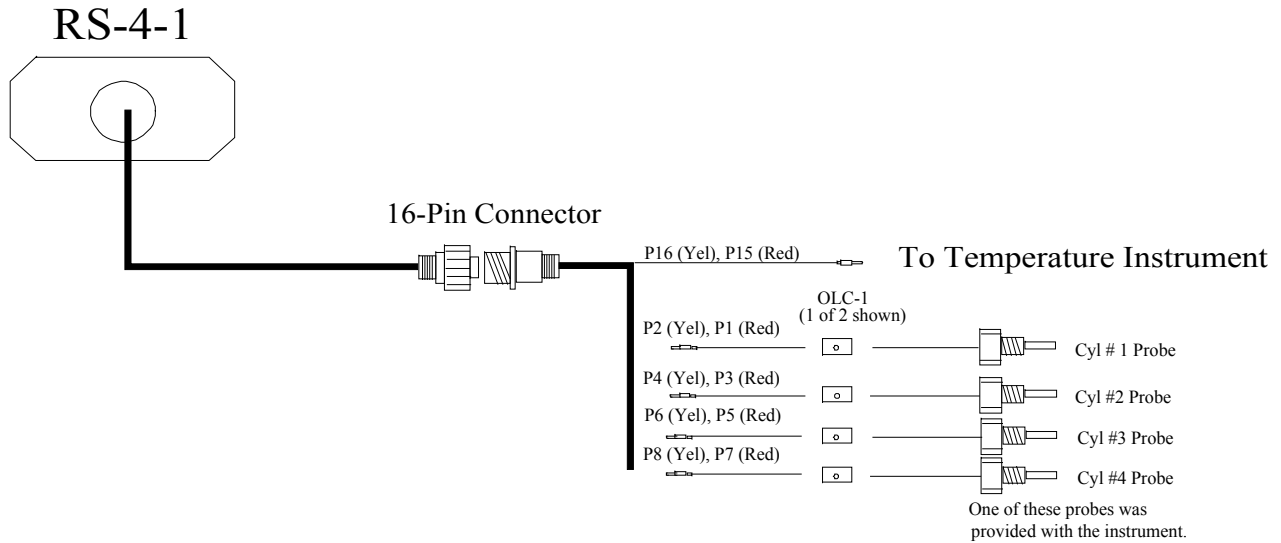
Instrument Wiring Diagram

1031121



Remote Switch Wiring Diagram

RS-4-1, RS-4-2, RS-6-1 and RS-6-2

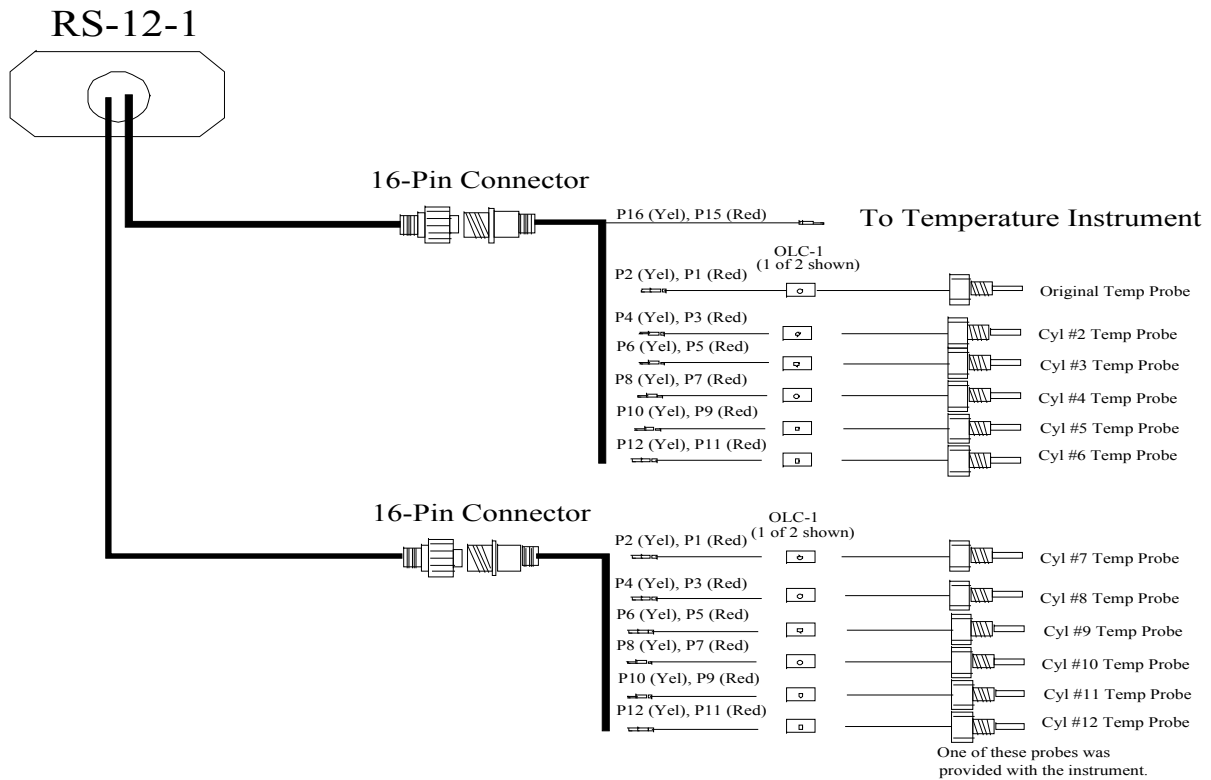
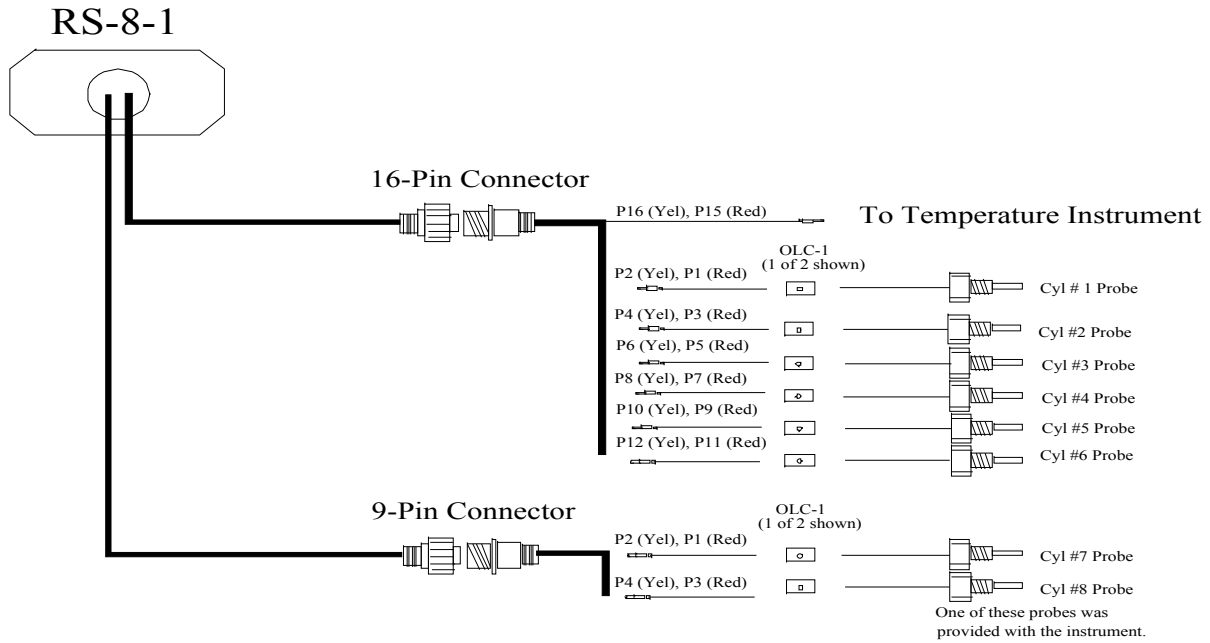


RS-4-2 and RS-6-2 Only

From the back of the RS-6-2 and RS-4-2 there are two groups of the above wires. Connect one group to the EGT Probes (or left channel for a twin installation) and connect the other group to the CHT (or right channel for a twin installation).

Remote Switch Wiring Diagram

RS-8-1 and RS-12-1



United States of America
Department of Transportation — Federal Aviation Administration
Supplemental Type Certificate

Number SA1626NM

This certificate, issued to Electronics International, Inc.

*certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part * of the Civil/Federal Regulations.*

Original Product — Type Certificate Number: * See attached Approved Model List No. SA1626NM
Make: for list of approved airplane models and
Model: applicable airworthiness regulations.

Description of Type Design Change: Installation of Electronics International EGT's/CHT's, remote switches, probes, cables, and accessories in accordance with Electronics International, Inc. Installation Instruction No. 070781, dated February 25, 1985, or later FAA approved revisions.

| <u>Model</u> | <u>Type</u> | <u>Model</u> | <u>Type</u> |
|--------------|---------------|--------------|------------------------|
| EC-1 | EGT/CHT | C-6 | 6 Channel CHT |
| E-4 | 4 Channel EGT | EC-2 | EGT/CHT |
| E-6 | 6 Channel EGT | E-2 | EGT |
| C-4 | 4 Channel CHT | C-2 | CHT (Continued page 3) |

Limitations and Conditions: Approval of this change applies to the above model aircraft only. This approval should not be extended to aircraft of this model on which other previously approved modifications are incorporated unless it is determined that the relationship between this change and any of those other previously approved modifications, including changes in type design will introduce no adverse effect upon the airworthiness of that aircraft. A copy of this Certificate, Addendum, Approved Model List No. SA1626NM, and Electronics International, Inc. Installation Instructions No. 070781, dated February 25, 1985, or later FAA approved revisions, must be maintained as part of the permanent records for the modified aircraft. (Cont. pg. 3)

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: May 30, 1982

Date issued:

Date of issuance: September 20, 1982

Date amended: April 13, 1983; May 7, 1986;
July 30, 1992

By direction of the Administrator



(Signature)
Acting Manager, Special Certification Branch
Seattle Aircraft Certification Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

United States of America
Department of Transportation—Federal Aviation Administration
Supplemental Type Certificate
(Continuation Sheet)

Number SA1626NM

SUPPLEMENTAL TYPE CERTIFICATE ADDENDUM NO. SA1626NM

Description of Type Design Change: (Continued from page 1)

| <u>Model</u> | <u>Type</u> |
|--------------|--|
| RS4-1 | 4 Channels, Single Function |
| RS-6-1 | 6 Channels, Single Function |
| RS-8-1 | 8 Channels, Single Function |
| RS-12-1 | 12 Channels, Single Function |
| RS-4-2 | 4 Channels, Dual Function |
| RS-6-2 | 6 Channels, Dual Function |
| E-1 | Single Channel, EGT Unit |
| EC-1D | Single Channel, EGT Unit with Differential |
| C-1 | Single Channel, CHT Unit |

Limitations and Conditions: (Continued)

NOTE: The Electronic International EGT's/CHT's monitoring units are not approved for use as alternates to required equipment. If the EGT/CHT gauge is a required instrument on the aircraft, then Electronic International's monitoring units can be used, only if it is a standby to the primary instrument.

- END -

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

List of Active Pages
for
Approved Model List (AML) SA1626NM

3/9/93

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| 5 ----- | 3/9/93 |
| 6 ----- | 6/2/92 |
| 7 ----- | 6/2/92 |
| 8 ----- | 6/2/92 |

FAA Approved by: _____

Samuel L. King

Date: _____

4-6-93

FAA Approved
Electronics International, Inc.
Approved Model List (AML) SA1626NM
for

Installation Of Electronics International Digital EGT/CHT Instruments and Accessories

Issue Date: May 24, 1988

| Item | Aircraft Make | Aircraft Model | Original Type Certificate Number | Certification Basis for Alteration | FAA Sealed Drawings | | Installation Instructions | | AML Amend. Date |
|------|------------------------------------|--|---|---|---------------------|----------|---------------------------|---------|--------------------------------|
| | | | | | Number | Rev. No. | Number | Date | |
| | | | | | TD121161 | 12/11/86 | 070781 | 2/25/85 | |
| 1. | Aerocar Inc. | I | 4A16 | CAR 3 | " | " | " | " | None |
| 2. | Aeronca Inc. (also see trytek) | C-2 Series C-3 Series K&S0 Series L Series 15 Series | ATC 351 A 396 -688, -676 ATC 596 & 614 A802 | FAR 23 " " " | " | " | " | " | 6/02/92 |
| 3. | Aero-spatiale (Also see Socata) | SE 3160 Alovette-III SA 315 Series SA 316 Series SA 319 Series 262A Series SN-601 SA-350 & 360 Series SA-365 Series | H1 IN H1 IN H1 IN H1 IN H1 IN A6 EU A37 EU H8 EU H10 EU | CAR 10 CAR 10 CAR 10 CAR 10 CAR 10 CAR 10 FAR 21.29 FAR 21.29 FAR 21.29 | " | " | " | " | 6/02/92 " " " |
| 4. | Aircraft Associates Inc. | J-2 | ATC 620 | FAR 23 | " | " | " | " | 6/02/92 |
| 5. | Air Tractor Inc. | AT-300, -300, -302, -400 | A9SW | FAR 21.25 (a)(1) | TD121161 | | | | 5/24/88 |
| 6. | Artic Aircraft Co. | S-1A Series S-1B Series | A737 A754 | CAR 4a CAR 4a | " | " | " | " | None |
| 7. | Ayres Corp. | S-2R S Series 600 Series S2R Series | A3SW A4SW 2A9 2A7 A3SW A4SW | CAR 3 CAR 8 CAR 8.10(a)(1) CAR 3 CAR 8 | " | " | " | " | None 6/02/92 " " " |
| 8. | Ballauer | Funk Series | 2-564 | CAR 04.031 | " | " | " | " | None |
| 9. | Balloon Works, The | Fire Flys | A14S0 | FAR 31 | " | " | " | " | 3/09/93 |

FAA Approved
Electronics International, Inc.
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Installation Of Electronics International Digital EGT/CHT Instruments and Accessories

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| | | | | | Number | Rev. No. | Number | Date | |
| | | | | | TD121161 | 12/11/86 | 070781 | 2/25/85 | |
| 10. | Beech Aircraft Corp. | B17B,B17L, S817L B17R C17B,C17L C17R 18 Series 19,23,24 Series 33,35,36 Series 45 Series 50 Series 60 Series 65, 70, 90 Series 76 Series 77 Series F90 Series 95,55,56, 58 Series 99, 100 Series 200 Series 300,1900,& 1900C Series | ATC 560 ATC 579 ATC 602 ATC 604 A-757, -765, A1CE 3A15 5A3 5A4 A12CE 3A20 A29CE A30CE A31CE 3A16 A23CE A14CE A24CE A24CE | CAR 4a CAR 4a CAR Part 3 CAR Part 3 Car Part 03 CAR Part 3 FAR Part 23 CAR 3 Part 23 Part 23 Part 23 CAR Part 3 FAR Part 23 FAR Part 23 FAR Part 23 FAR Part 23 | " | " | " | " | 3/09/93 " 6/02/92 6/02/92 None 6/02/92 None 6/02/92 " 6/02/92 |
| 11. | Bell Helicopter Textron | 47 Series 204&205 Series 212 Series 214 Series 222 Series 206 Series | H-1, 2H3, 2-H1 H1SW H4SW H6SW H9SW H2SW | CAR 6 CAR 7 FAR Part 29 FAR Part 29 FAR Part 29 CAR 6 | " | " | " | " | 6/02/92 " " |
| 12. | Bellanca Aircraft Corp. | 7 Series 11 Series 14, 17, Series 8 Series 300 Series 400 Series | A-759, A-761, A-18CE A21CE ATC328 ATC319 | CAR 4a CAR 4a FAR Part 23 FAR Part 23 FAR Part 23 FAR Part 23 | " | " | " | " | None " |
| 13. | British Aerospace | Twin Pioneer Series 2&3 Beagle B206 Series 1&2 Beagle B121 Series 1,2, &3 | 7A7 A11EU A22EU A22CE | CAR 10 FAR 21.29 (CAR 3) FAR 21.29 (FAR 23) | " | " | " | " | None |
| 14. | Camair Aircraft Corp. | Navion 480 | 2A2 | CAR 3 | " | " | " | " | 6/02/92 |

Installation Of Electronics International Digital EGT/CHT Instruments and Accessories

| Item | Aircraft Make | Aircraft Model | Original Type Certificate Number | Certification Basis for Alteration | FAA Sealed Drawings | | Installation Instructions | | AML Amend. Date |
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| | | | | | Number | Rev. No. | Number | Date | |
| | | | | | TD121161 | 12/11/86 | 070781 | 2/25/85 | |
| 15. | Cessna Aircraft (Also see Reims) | 120,140 Series 150, 152 Series 170 Series 172,175 Series 177 Series 180 Series 182, 185 Series 188 Series 190,195 Series 206 Series 207 Series 210 Series 303, 305 Series 310 Series 320, 335, 340 Series 336, 337, 401, 402, 411, 414, 421, 425 Series 404 | A768, 5A2 3A19 A799 3A12, 3A17 A13CE,A20CE 5A6 3A13, 3A24 A-761 A9CE A-790 A4CE A16CE 3A21 5A5, 3A14 3A10 3A25 A2CE A6CE A7CE A25CE | CAR 4a CAR 3(CAR 4a) CAR Part 3 CAR Part 03 CAR Part 3 FAR Part 23 CAR Part 3 CAR Part 3 CAR 4a FAR Part 23 CAR 3 CAR Part 3 FAR Part 23 CAR Part 3 CAR Part 3 CAR 3 CAR Part 3 CAR 3 CAR Part 3 CAR Part 3 FAR Part 23 | " | " | " | " Rev B 5/23/86 2/25/85 | None 03/09/93 6/02/92 6/02/92 None " |
| 16. | Christen Industries Inc. | S-1,S-2, Series (Pitts) | A8S0 | FAR 21 (FAR 23) | | " | " | " | None |
| 17. | Child Doyle F. | S-1, S-2 Series (Pitts) | A8S0 | FAR 21 (FAR 23) | " | " | " | " | 6/02/92 |
| 18. | Consolidated Aeronautic | Colonial C. Series, Lake LA Series | 1A13 | CAR 03 | " | " | " | " | None |
| 19. | De Havilland | DH Series DHC Series | A8EU 2-439 2-393 7A10 A-816 A-807 A9EA AR-33 AR-13 1A19 | FAR 21.29 FAR 21.29 FAR 21.29 CAR 10 CAR 10 CAR 10 CAR 3 CAR 8 CAR 8 CAR 10 | " | " | " | " | 6/02/92 " " " |
| 20. | Enstrom Helicopters | F Series 280 Series | HICE | CAR Part 6 | " | " | " | " | None |
| 21. | Great Lakes | 2T Series | ATC354 ATC-167 2-3339 ATC228 A18EA | FAR 23 | " | " | " | " | 6/2/92 |

FAA Approved
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| Item | Aircraft Make | Aircraft Model | Original Type Certificate Number | Certification Basis for Alteration | FAA Sealed Drawings | | Installation Instructions | | AML Amend. Date |
|------|---|--|---|---|---------------------|----------|---------------------------|---------|------------------|
| | | | | | Number | Rev. No. | Number | Date | |
| | | | | | TD121161 | 12/11/86 | 070781 | 2/25/85 | |
| 22. | Gulfstream Aerospace | 111 Series 112 Series 114 Series | 11SO A12SO A12SO | FAR 23 (FAR 36) | " | " | " | " | 6/02/92 |
| 23. | Gulfstream American Commander Division | Gumman: G-164 Series | 1A16 | CAR 8.10 (a)(1) CAR 4b FAR 23 | " | " | " | " | None |
| | | G Series AA Series GA Series 500 Series 111 112Series, 114Series | 1A17, A12EA A11EA A16EA A17SO 6A1 A11SO A12SO | FAR 23 CAR 3 FAR 23 (Far 21) FAR 23 (Far 36) | " | " | " | " | 6/02/92 " |
| 24. | Helio Aircraft LTD (Navion) | H-250 H-295 HT-295 H-391 H-391B H-395 H-395A 500 | 1A8 1A8 1A8 1A8 1A8 1A8 A2EA | CAR 3 CAR 3 CAR 3 CAR 3 CAR 3 CAR 3 CAR 3 | " | " | " | " | None |
| 25. | Hiller Aviation | UH-12 Series | 6H1, 6H2, 4H10, 4H11, H1WE | CAR 6 | " | " | " | " | None |
| 26. | Huges Helicopters | 300 Series 269 Series | 4H12 | CAR Part 6 | " | " | " | " | 6/02/92 None |
| 27. | Husky | A-1 | A22NM | FAR 23 | " | " | " | " | 6/02/92 |
| 28. | Hynes | H-2, H-4 | 2H2 | CAR Part 6 | " | " | " | " | None |
| 29. | Industrie Aeronautiche E Meccaniche (See Piaggio) | PD 808/526 P. 166 Series | A12EU 7A4 | FAR 21.29 CAR 10 | | | | | 3/09/93 " |
| 30. | Lake | See consolidated Aeronautics | | | | | | | |
| 31. | Maule Aircraft Corp. | M Series | 3A23 | CAR Part 3 | " | " | " | " | None |
| 32. | McDonnell Douglas | DC3 Series DC4 Series | A-807 A-618 | CAR 7a CAR 7a | " | " | " | " | 6/02/92 |
| 33. | Mooney | M20 Series (Mooney) M22 | 2A3 A6SW | CAR 3 CAR 3 | " | " | " | " | None 6/02/92 |
| 34. | Navion | See Caniar Ryan and Thompson | | | | | | | |

Installation Of Electronics International Digital EGT/CHT Instruments and Accessories

[illegible]

FAA Approved
Electronics International, Inc.
Approved Model List (AML) SA1626NM
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Installation Of Electronics International Digital EGT/CHT Instruments and Accessories

Issue Date: May 24, 1988

| Item | Aircraft Make | Aircraft Model | Original Type Certificate Number | Certification Basis for Alteration | FAA Sealed Drawings | | Installation Instructions | | AML Amend. Date |
|------|---|---|---|---|---------------------|----------|---------------------------|---------|---------------------------|
| | | | | | Number | Rev. No. | Number | Date | |
| | | | | | TD121161 | 12/11/86 | 070781 | 2/25/85 | |
| 42. | Reims Aviation (Cessna) | 172Series 150Series 182Series 337Series 177Series | A4EU A18EU A13EU A42EU A23EU A26EU | CAR 10 FAR 21.29 (CAR 3) FAR 21.29 CAR PART 3 FAR 21.29 (FAR 23) FAR PART 23 | " | " | " | " | None |
| 43. | Robinson Helicopter | R 22 | H10WE | FAR 27 | " | " | " | " | None |
| 44. | Rockwell International (Also see Ayres) | 111Series 112Series 114Series 500,520 560,680, 681,690, 695,720 | A12S0 A12S0 A12S0 A7S0, 6A1 | FAR 23 (FAR 36) FAR 23, CAR 3 | " | " | " | " | None 6/02/92 |
| 45. | Rust, Robert F. (Chipmunk) | 21,22 Series | A44EU | FAR 21.29 | " | " | " | " | None |
| 46. | Ryan Aeronautical Co. | SCW-145 ST Series | 658 ATC 571 ATC 681 A-749 | CAR 4a CAR 4a | " | " | " | " | 6/02/92 |
| 47. | Ryan Aircraft Corp. | B Series | ATC 25 | CAR 4a | " | " | " | " | 6/02/92 |
| 48. | Sikorsky Aircraft | H19 Series S-39Series S-41-B S-43Series VS-44-A S-51Series S-52Series S-55Series S-58Series S-61Series S-62A Series S-64Series S-76Series | HR1S0 ATC-375 2-286 A-593 752 H-2 H-3 1H4 1 H11 1 H15 1 H13 H1EA H1NE | FAR 21.25 FAR 21.25 FAR 21.25 Aero Bul-7A CAR 4A CAR 6 CAR 6 CAR 6 CAR 6 CAR 7 CAR 7 CAR 8 FAR 29 | " | " | " | " | 6/02/92 " " |
| 49. | Socata Group Aero-spatiale | TB9,10, 20,21 | A51EU | | " | " | " | " | 6/02/92 |
| 50. | Stearman Aircraft | C3 Series 4 Series 6 Series | ATC55, 2-159 2-445 ATC251 2-155 ATC-305 ATC-292 ATC459 | FAR 23 FAR 23 FAR 23 | " | " | " | " | 6/02/92 " " |

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Issue Date: May 24, 1988

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|------|---------------------------------------|--|---|---|---------------------|----------|---------------------------|---------|-----------------|
| | | | | | Number | Rev. No. | Number | Date | |
| | | | | | TD121161 | 12/11/86 | 070781 | 2/25/85 | |
| 51. | Stinson | SM Series | ATC16 ATC136 ATC48 ATC145 ATC161 ATC294 ATC98 | CAR 4a | " | " | " | " | 6/02/92 |
| | | SR Series | ATC519 ATC530 ATC580 ATC594 ATC608 ATC609 | CAR 4a | | | | | " |
| | | SR Series | ATC621 ATC625 ATC640 | CAR 4a | | | | | " |
| | | | | | | | | | " |
| 52. | Stol Amphibian Corp. | UC-1 (Twin-Bee) Republic RC-3 | A6EA | CAR 3 | " | " | " | " | None |
| | | | A-769 | CAR 03 | " | " | " | " | " |
| 53. | Swift Museum Foundation | GC Series | A-766 | CAR 4a | | " | " | " | 6/02/92 |
| 54. | Taylorcraft Aviation | BC & BCS Series 19 & F21 DC Series BF Series BL Series | A-696 1A9 A746 A-699 A-700 | CAR Part 04 CAR Part 3 CAR 4a CAR 4a CAR 4a | " | " | " | " | None |
| 55. | Thompson, Jimmie, Enterprise (Navion) | A through Z Series | A-782 | CAR 3 | " | " | " | " | 6/02/92 |
| 56. | Trytek, E.J. | Aeronca CF, KC K, KS | ATC 655 A-634 | FAR 23 FAR 23 | " | " | " | " | 6/02/92 |
| 57. | Univar Aircraft Corp. (Forney) | 108 Series | A-767 A-718 | CAR 3 | " | " | " | " | None |
| | | F-1 | | CAR 3 | | | | | |
| | | F-1A | A-787 | CAR 3 | | | | | |
| | | E | A-787 | CAR 3 | | | | | |
| | (Erco) (Stinson) | (Mooney) | | CAR 3 | | | | | |
| | | G | A-787 | CAR 3 | | | | | |
| | | 415-D | A-787 | CAR 3 | | | | | |
| | | A-2 | A-787 | CAR 3 | | | | | |
| | (Alon) | A2-A | A-787 | CAR 3 | | | | | |
| | | M10 | A-787 | CAR 3 | | | | | |
| | | 415-C | A-718 | CAR 4a | | | | | |
| | | 415-CD | A-718 | CAR 4a | | | | | |
| 58. | Varga Aircraft Corp. | 2150Series | 4A19 | CAR 3 | " | " | " | " | None |

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|------|--------------------|----------------|--|------------------------------------|---------------------|----------|---------------------------|---------|-----------------|
| | | | | | Number | Rev. No. | Number | Date | |
| | | | | | TD121161 | 12/11/86 | 070781 | 2/25/85 | |
| 59. | Waco Aircraft | A Series | ATC26, 644 677 598 714 ATC41 ATC123 | CAR 4a | " | " | " | " | 6/02/92 |
| | | B&C Series | ATC168 ATC538 | " | " | " | " | " | " |
| | | D Series | ATC639 ATC597 ATC42 | " | " | " | " | " | " |
| | | E Series | 665 2-430 | " | " | " | " | " | " |
| | | G Series | ATC 13 2-363 | " | " | " | " | " | " |
| | | I, J, K Series | 2-361 ATC 313 | " | " | " | " | " | " |
| | | O, P, Q Series | ATC468 ATC464 ATC491 ATC453 ATC502 | " | " | " | " | " | " |
| | | R&S Series | ATC466 ATC311 ATC543 | " | " | " | " | " | " |
| | | U, Y, Z Series | ATC542 ATC586 ATC627 | " | " | " | " | " | " |
| | | | | | " | " | " | " | " |
| 60. | Weatherly Aviation | 201Series | A10WE | FAR 21.25 (a) (CAM8) | " | " | " | " | None |
| | | 620Series | A26WE | FAR 21.25 (a) | " | " | " | " | None |
| | | | | | | | | | |

United States of America
Department of Transportation — Federal Aviation Administration
Supplemental Type Certificate

Number SA2350NM

This certificate, issued to Electronics International, Inc.

*certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part * of the Civil Air Regulations.*

Original Product — Type Certificate Number: * * See Attached Master Eligibility List
Make: * (MEL) Number SA2350NM for list of
Model: * approved airplane models and applicable
airworthiness regulations.

Description of Type Design Change: Installation of Electronics International, Inc. Digital Carburetor/Outside Air Temperature Gauges listed on the attached Continuation Sheet, and the associated accessories in accordance with Electronics International, Inc. Installation Instructions No. II 070781-1.

Limitations and Conditions:

Approval of this change in type design applies to the above referenced models only. This approval should not be extended to other aircraft of these models on which other previously approved modifications are incorporated unless it is determined that the relationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that aircraft. (Continued on Page 3 of 3, Continuation Sheet.)

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: July 8, 1983

Date reissued:

Date of issuance: February 16, 1984

Date amended: May 26, 1987



By direction of the Administrator

D. L. Ruggin
(Signature)

Assistant Manager, Seattle
Aircraft Certification Office

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

United States of America
Department of Transportation—Federal Aviation Administration
Supplemental Type Certificate
(Continuation Sheet)

Number
SUPPLEMENTAL TYPE CERTIFICATE NO. SA2350NM

Description of Type Design Change: (Continued)

Approved Electronics International, Inc. Carburetor/Outside Air Temperature Gauges:

| <u>Model</u> | <u>Type</u> |
|--------------|------------------------------|
| EAE-1 | EGT Left, OAT and EGT Right |
| A-1 | OAT Gauge |
| EAC-1 | EGT, OAT and CHT Gauge |
| CA-1 | Carb/Outside Air Temp. Gauge |

NOTE: The models listed herein are designed for monitoring exhaust gas and cylinder head temperatures and are not primary instruments.

Limitations and Conditions: (Continued)

A copy of this Certificate, Continuation Sheet, Master Eligibility List (MEL) SA2350NM, Electronics International, Inc. Installation Instructions No. II 070781-1 and Operating Instructions No. OI 040831 shall be maintained as part of the permanent records for the modified aircraft.

- END -

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

LIST of ACTIVE PAGES
for
Approved Model List (AML) SA2350NM

DATE
June 6, 1990

| <u>PAGE</u> | <u>AMENDMENT DATE</u> |
|-----------------|-----------------------|
| This Page ----- | June 6, 1990 |
| 1 ----- | None |
| 2 ----- | None |
| 3 ----- | None |
| 4 ----- | June 6, 1990 |
| 5 ----- | None |
| 6 ----- | None |
| 7 ----- | None |
| 8 ----- | None |
| 9 ----- | None |
| 10 ----- | None |
| 11 ----- | June 6, 1990 |

FAA Approved Stewart K Miller

Amended _____

FAA APPROVED

ELECTRONICS INTERNATIONAL, INC.

MASTER ELIGIBILITY LIST (MEL) SA2350NM

FOR

INSTALLATION OF ELECTRONICS INTERNATIONAL DIGITAL CARBURETOR/OUTSIDE AIR TEMPERATURE GAUGES

DATE: February 16, 1984

| ITEM | AIRCRAFT MAKE | AIRCRAFT MODEL | ORIGINAL TYPE CERTIFICATE NUMBER | CERTIFICATION BASIS FOR ALTERATION | FAA SEALED DRAWING/ DRAWING LIST | | AFM SUPPLEMENT NUMBER/ DATE | MEL AMENDMENT DATE |
|------|--------------------------------|---|--|--|--|------|--------------------------------------|--------------------------|
| | | | | | NUMBER | REV. | | |
| 1. | Aerocar Inc. | I | 4A16 | CAR 3 | TD 010972 | | | |
| 2. | Aeronca Inc. | C-2 Series C-3 Series K & 50 Series L Series 15 Series | ATC 351 A 396 688, 676 ATC 596 & 614 A802 | ATC 351 Aero Bul. 7-A 688, 676 ATC 596 & 614 CAR 3 | | | | |
| 3. | Aircraft Associates Inc. | J-2 | ATC 620 | ATC 620 | | | | |
| 4. | Arctic Air- craft Co. | S-1A Series S-1B Series | A737 A754 | CAR 4a CAR 4a | | | | |
| 5. | Ayres Corp. | S2R Series | A3SW, A4SW | CAR 3 CAR 8 | | | | |
| 6. | Air Tractor Inc. | AT-300, -300, -302, -400 | A9SW | FAR 21.25(a)(1) | | | | |
| 7. | Ballouer | Funk Series | 2-564 | CAR 04.031 | | | | |

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|------|---------------------------------|--|---|--|--|--------------------------|----------------------------------|--------------------------|
| | | | | | NUMBER | REVISION NO. AND DATE | | |
| *8. | Beech Air- craft Corp | 18 Series 19, 23, 24 Series 33, 35, 36 Series 45 Series 50 Series 60 Series 65, 90 Ser. 76 Series 77 Series 95, 55, 56 Series 58 Series Series | A-757, -765 A-1CE 3A15 5A3 5A4 A12CE 3A20 A29CE A30CE 3A16 A23CE | CAR 4a CAR 03 CAR Part 3 CAR Part 3 CAR Part 03 CAR Part 3 FAR Part 23 CAR 3 Part 23 Part 23 CAR Part 3 FAR Part 23 | TD 010972 | | | |
| 9. | Bell Heli- copter Textron | 47 Series | H-1, 2H3, 2H1 | CAR 6 | TD 010972 | | | |

* NOTE: Amended Item #8 to remove Beech F90, 99, 100 and 200 Series aircraft on 5/26/87.

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|------|-------------------------------|--|---|--|--|------|--------------------------------------|--------------------------|
| | | | | | NUMBER | REV. | | |
| 10. | Bellanca Aircraft Corp. | 7 Series 11 Series 14, 17 Series 8 Series 300 Series 400 Series | A-759 A-761, 796 A-18CE A21CE ATC328 ATC319 | CAR Part 4a CAR 4a CAR 3 FAR Part 23 FAR Part 23 ATC328 ATC319 | TD 010972 | | | |
| 11. | British Aerospace | Twin Pioneer Series 2 & 3 Beagle B206 Series 1 & 2 Beagle B121 Series 1, 2 & 3 | 7A7 A11EU A22EU | CAR 10 FAR 21.29 (CAR 3) FAR 21.29 (FAR 23) | | | | |
| 12. | Cessna Aircraft | 120, 140 Series 150, 152 Series 170 Series 172, 175 Series 177 Series 180 Series | A768, 5A2 3A19 A799 3A12, 3A17 A13CE, A20CE 5A6 | CAR 4a CAR 3 (CAR 4a) CAR Part 3 CAR Part 03 CAR Part 3 FAR Part 23 CAR Part 3 | | | | |

INSTALLATION OF ELECTRONICS INTERNATIONAL DIGITAL CARBURETOR/OUTSIDE AIR TEMPERATURE GAUGES

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|------|---------------------------------------|--|--|--|---|--------------------------|-----------------------------------|--------------------------|
| | | | | | NUMBER | REVISION NO. AND DATE | | |
| 12 | Cessna Aircraft (Cont.) | 182, 185 Series 188 Series 190 Series 206 Series 207 Series 210 Series 303, 305 Series 310 Series 320,335,340 Series 336, 337 401, 402 411,414,421 425 Series 404 Series | 3A13, 3A24 A9CE A-790 A4CE A16CE 3A21 5A5, 3A14 3A10 3A25 A2CE, A6CE A7CE A25CE | CAR Part 3 FAR Part 21 CAR 3 CAR Part 3 FAR Part 23 CAR Part 3 CAR Part 3 CAR 3 CAR Part 3 CAR 3, CAR Part 3 CAR Part 3 FAR Part 23 | TD 010972 | | | 6/06/90 |
| 13 | Child Doyle F. | S-1, S-2, Series (Pitts) | A8S0 | FAR 21 (FAR 23) | | | | |
| 14 | Consoli- dated Aero- nautics | Colonial C. Series, Lake LA Series | 1A13 | CAR 03 | | | | |
| 15 | Enstrom Helicopter | F Series, 280 Series | HICE | CAR Part 6 | | | | |

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| | | | | | NUMBER | REV. | | |
| 16. | Great Lakes | 2T Series | ATC-167 2-3339 ATC228 ATC354 A18EA | ATC-167 2-3339 ATC228 ATC354 FAR 23 | TD 010972 | | | |
| 17. | Gulfstream American | Grumman: G-164 Series G Series AA Series GA Series | 1A16 1A17, A12EA A11EA, A16EA A17SO | CAR 8.10(a)(1) CAR 4b FAR 23 FAR 23 | | | | |
| 18. | Hiller Aviation | UH-12 Series | 6H1, 6H2, 4H10, 4H11, HIWE | CAR 6 | | | | |
| 19. | Hughes Helicopters | 300 Series, 269 Series | 4H12 | CAR Part 6 | | | | |
| 20. | Hynes | H-2, H-4 Series | 2H2 | CAR Part 6 | | | | |
| 21. | Maule Air- craft Corp. | M Series | 3A23 | CAR Part 3 | | | | |
| 22. | Mooney Aircraft | M20 Series | 2A3 | CAR 3 | | | | |

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| | | | | | NUMBER | REV. | | |
| 23. | North American Aviation | AT-6 Series | A-2-575 | CAR 4a | TD 010972 | | | |
| 24. | Partenavia Costruzioni Aeronautiche | P68 Series | A31EU | FAR 21.29 (FAR 23) | | | | |
| 25. | Pilatus Britten- Norman | BN Series | A17EU | FAR 21.29, FAR 23 | | | | |
| 26. | Piper Air- craft Corp. | J2 Series J3 Series J4 Series J5 Series PA12 Series PA14 Series PA15 Series PA16 Series PA17 Series PA18 Series PA20 Series | ATC595 ATC660, A-691, A-692, A698, 695 A-703, 708, A-740, 721 A-725 A-780 A-797 A-800 1A1 A-805 1A2, AR-7 1A4 | ATC595 ATC 660 CAR 4a CAR 4a CAR 4a 695 CAR 4a 708 CAR 4a CAR 4a CAR 3 CAR 3 CAR 3 CAR 3 CAR 3 CAR 3 CAR 3 CAR 8.10(b) CAR 3 | | | | |

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| | | | | | NUMBER | REV. | | |
| 26. | Piper Air- craft Corp. (Cont'd) | PA22 Series | 1A6 | CAR 3 | TD 010972 | | | |
| | | PA23 Series | 1A10 | CAR 3 | | | | |
| | | PA24 Series | 1A15 | CAR 3 | | | | |
| | | PA25 Series | 2A8, 2A10 | CAR 3 CAR 8.10(b) | | | | |
| | | PA 30, 39, 40 Series | A1EA | CAR 3 | | | | |
| | | PA31 Series | A8EA | CAR 3 (FAR 23) | | | | |
| | | PA36 Series | A9SO, A10SO | FAR 23 FAR 21 | | | | |
| | | PA38 Series | A18SO | FAR 23 | | | | |
| | | PA28 Series | 2A13 | CAR 3 | | | | |
| | | PA32 Series | A3SO, A7SO | CAR 3 FAR 23 | | | | |
| | | PA44 Series | A19SO | FAR 23 | | | | |
| 27. | Prop-Jets Inc (Aero Commander) | 200 Series | 3A18 | CAR 3 | | | | |
| 28. | Reims Avia- tion (Cessna) | 172 Series | A4EU, A18EU | CAR 10 FAR 21.29 (CAR 3) | | | | |
| | | 150 Series | A13EU | FAR 21.29 | | | | |
| | | 182 Series | A42EU | CAR Part 3 | | | | |
| | | 337 Series | A23EU | FAR 21.29 (FAR 23) | | | | |
| | | 177 Series | A26EU | FAR Part 23 | | | | |
| 29. | Robinson Helicopter | R22 | H10WE | FAR 27 | | | | |

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| | | | | | NUMBER | REV. | | |
| 30. | Rockwell International | 111 Series, 112 Series, 11 Series | A12S0 | FAR 23 (FAR 36) | TD 010972 | | | |
| 31. | Rust, Robert E. (Chipmunk) | 21, 22 Series | A44EU | FAR 21.29 | | | | |
| 32. | Stearman Aircraft | C3 Series | ATC55, 2-159 2-445 ATC251 | ATC55 2-159 2-445 ATC251 | | | | |
| | | 4 Series | 2-155 ATC305 ATC292 | 2-155 ATC305 ATC292 | | | | |
| | | 6 Series | ATC459 | ATC459 | | | | |
| 33. | Stinson Division | SM Series | ATC16 ATC136 ATC48 ATC145 ATC161 ATC194 ATC298 | ATC16 ATC136 ATC48 ATC145 ATC161 ATC194 ATC298 | | | | |
| | | SR Series | ATC519 ATC530 ATC580 ATC594 ATC608 ATC609 | ATC519 ATC530 ATC580 ATC594 ATC608 ATC609 | | | | |

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| | | | | | NUMBER | REV. | | |
| 33. | Stinson Division (Cont'd) | SR Series (Cont'd) | ATC621 ATC625 ATC640 | ATC621 ATC625 ATC640 | TD 010972 | | | |
| 34. | Taylorcraft Aviation | BC & BCS Series 19 & F21 Series DC Series BF Series BL Series | A-696 1A9 A746 A-699 A-700 | CAR Part 04 CAR Part 3 CAR 4a CAR 4a CAR 4a | | | | |
| 35. | Thompson, Jimmie, Enterprise (Navion) | A through H Series | A-782 | CAR 3 | | | | |
| 36. | Varga Air- craft Corp. | 2150 Series | 4A19 | CAR 3 | | | | |
| 37. | Waco Aircraft | A_ Series B & C Series | ATC 26, 664, 677, 598, 714, ATC 41 ATC 123 ATC 168, ATC 538 | ATC 26 664 677 598 CAR 4a ATC 41 ATC 123 ATC 168 ATC 538 | | | | |

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|-----|-----------------------------|---|--|--|--|------|--------------------------------------|--------------------------|
| | | | | | NUMBER | REV. | | |
| 37. | Waco Air- craft (Cont'd) | B & C Series (Cont'd) D Series E Series G Series I, J, K Series O, P, Q Series R & S Series U, Y, Z Series | ATC 362 ATC 240 ATC 352 ATC 575 ATC639 ATC597 ATC42 665 2-430 ATC 13 2-363 ATC345 2-361 ATC 313 ATC468 ATC464 ATC491 ATC453 ATC502 ATC466 ATC311 ATC543 ATC479 ATC473 ATC467 ATC499 ATC511 ATC546 ATC568 ATC626 | ATC 362 ATC 240 ATC 352 ATC 575 ATC639 ATC597 ATC42 665 2-430 ATC 13 2-363 ATC345 2-361 ATC 313 ATC468 ATC464 ATC491 ATC453 ATC502 ATC466 ATC311 ATC543 ATC479 ATC473 ATC467 ATC499 ATC511 ATC546 ATC568 Aero Bul. 7A | TD 010972 | | | |

FAA APPROVED MODEL LIST (AML) NO. SA2350NM
ELECTRONICS INTERNATIONAL, INC.
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|------|--|---|--|--|--|--------------------------|-----------------------------------|--------------------------|
| | | | | | NUMBER | REVISION NO. AND DATE | | |
| 37 | Waco Aircraft (Cont.) | U, Y, Z Series | ATC542 ATC586 ATC 627 | ATC542 ATC586 ATC627 | TD 010972 | | | 11/13/86 |
| 38 | Weatherly Aviation | 201 Series 620 Series | A10WE A26WE | FAR 21.25(a) CAM 8) FAR 21.25(a) | TD 010972 | | | |
| 39 | Stol Amphibian Corp. | UC-1 (Twin-Bee) Republic RC-3 | A6EA A-769 | CAR 3 CAR 03 | TD 121161 TD 121161 | 12/11/86 12/11/86 | | 6/06/90 6/06/90 |
| 40 | Univair Aircraft Corp. (Erco) (Forney) (Erco) (Mooney) (Erco) | (Stinson) 108, 108-1, -2, -3, -5 G F-1 F-1A E 415-D M10 415-CD | A-767 A-787 A-787 A-787 A-787 A-787 A-787 A-718 | CAR 3 CAR 3 CAR 3 CAR 3 CAR 3 CAR 3 CAR 4a | | | | 6/06/90 |