

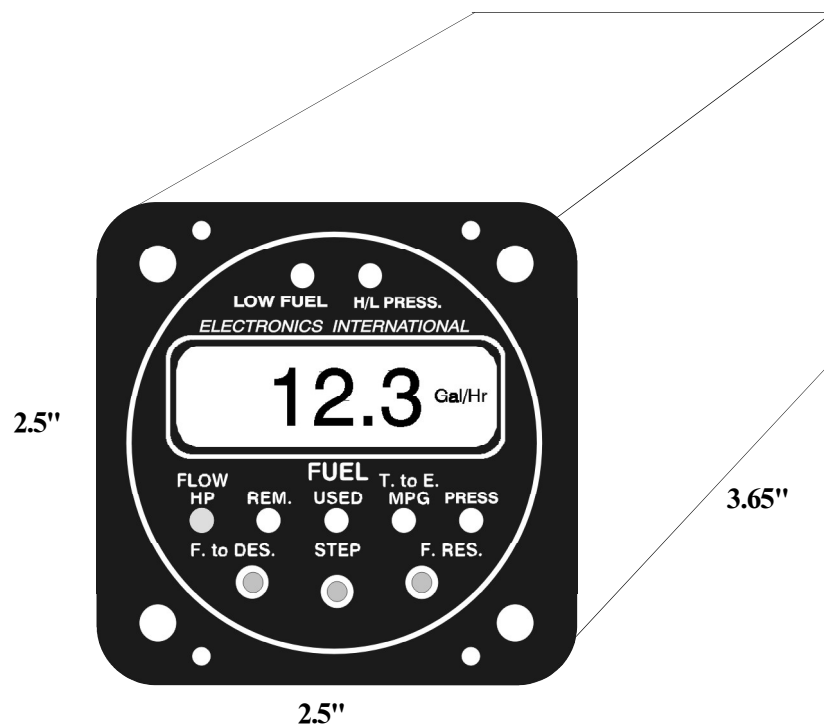
Primary Fuel Flow/Pressure ***(FP-5-LxxHyy and FP-5L-LxxHyy)*** ***Installation Instructions***

II S0506931

04/25/2001

Rev. C: 2/18/2005***

You must read this manual before installing or operating the instrument. This manual contains warranty and other information that may affect your decision to install this product and/or the safety of your aircraft.



Unit Model: _____ S/N: _____

Flow Transducer Model: _____ S/N: _____

Pressure Transducer Model: _____ S/N: _____



Electronics International Inc.®

63296 Powell Butte Hwy • Bend, OR 97701 • (541) 318-6060 • www.Buy-EL.com ☒

Important Notice

******* MUST READ *******

If you think it is not important to read this manual, you're wrong!
This manual contains important operating information that may affect the safety of you, your aircraft and passengers.

Read the Warranty / Agreement. There is information in the Warranty/Agreement that may alter your decision to install this product. **If you do not accept the terms of the Warranty / Agreement, do not install this product.** This product may be returned for a refund. Contact Electronics International inc. for details.

If you do not agree to and accept ALL the terms of this warranty, DO NOT Install This Product. You may return the product for a refund. Contact Electronics International Inc. for details.

By installing this product, the aircraft owner/pilot and installer agree to hold Electronics International Inc. harmless and in no way responsible for monetary compensation, including punitive damages for any incident, harm and/or damage associated with this product. If you do not agree to the above, **DO NOT INSTALL THIS PRODUCT.** This product may be returned for a refund. Contact Electronics International inc. for details.

The pilot **must** understand the operation and limitations of this product before flying the aircraft. Do not allow anyone to operate the aircraft that does not know how to properly interpret and operate this product. **Keep the Operating Manual in the aircraft at all times.** If you do not thoroughly understand the operation of this product, contact a knowledgeable flight instructor for training.

The ability for this product to respond to an engine or aircraft system anomaly is directly related to how that anomaly affects the reading of the function(s) being monitored (i.e.: if an engine fire does not affect the EGT or CHT, the EGT and CHT readings will not change).

This Instrument only displays the parameters for the function(s) being monitored. The pilot is responsible for interpreting the data and determining if an engine or aircraft system anomaly exists. When using this instrument, the pilot's diagnostic ability is limited to his/her interpretation of the displayed data and the there observation skills. To improve these skills the pilot should seek training from a flight instructor.

Check that the limit information on this instrument matches the published limits in your aircraft's P.O.H. or Flight Manual. This information may be listed in the T.C. Data Sheet for your aircraft. Any AD's and/or STC's may set forth additional limitations on the operation of your engine. The limit information listed in the AML is for unmodified aircraft and is intended for reference only. **It is the aircraft owner's and/or installer's responsibility to determine proper instrument calibration and range markings for your aircraft.**

The FT-60 is intended to be used on aircraft equipped with fuel pumps with engines rated below 350

H.P. A gravity feed fuel system or any engine rated over 350 H.P. *must* use an FT-90 flow transducer. An engine rated over 550 H.P. *must* use the FT-180 flow transducer.

Transducer Identification:

FT-60 - Red Cube
FT-90 - Gold Cube
FT-180 - Black Cube

If your aircraft is not covered on our STC (found at the back of this manual), you must perform the flow and pressure tests in FAA document A.C. 23-16 (Powerplant Guide for Certification of Part 23 Airplanes) to insure safe and proper operation.

Installation of the FP-5 on an aircraft with a fuel return line from the Pressure Carburetor requires a FFDM-1 Differential Module (see price sheet).

The placard "Do Not Rely on Fuel Flow Instrument to Determine Fuel Levels in Tanks" must be mounted on the aircraft instrument panel near the FP-5.

If after reading this manual you do not have the knowledge to interpret the displayed data to operate the aircraft safely or to detect engine and/or aircraft system problems, contact a knowledgeable instructor for training prior to flying the aircraft with this instrument.

If you detect a problem using this instrument, it is your responsibility to take appropriate action to insure the safety of the flight. Practice simulating problems to build your skills and to improve your understand of the relationships between problems and their affects on the displayed data. To insure you are taking appropriate action, contact a knowledgeable flight instructor for training.

This manual does not make any recommendations as to specific operating parameters or controlling methods. Check the airframe and/or engine manufacturer's recommendations to properly operate the aircraft systems and engine. It is the pilot's responsibility to operate the engine and aircraft safely.

It is possible for any instrument to fail thereby displaying inaccurate high, low or jumpy readings. Therefore, you **must** be able to recognize an instrument failure and you **must** be proficient in operating your aircraft safely in spite of an instrument failure. If you do not have this knowledge, contact the FAA or a knowledgeable flight instructor for training prior to flying the aircraft with this instrument.

Electronics International Inc. is not liable or responsible for a pilot's action or any situation that results in personal injury, property damage, missed commitments, lack of use of an aircraft or any expenses incurred due to: product failure, inaccuracy in displayed data or text files, display or display format issues, software bugs or problems, upgrade or customization issues, misinterpretation of the display, warning and/or limit settings, calibration problems, installation issues (leaks, mis-wiring, obstructions, damage to aircraft or components, incorrect installation of any parts, wrong parts, parts that don't fit, etc.) or any other issues related to the installation or operation of this product. All of the above are solely the pilot's and/or installer's responsibility. The pilot **must** understand the operation of this product before flying the aircraft. The pilot will not allow anyone to operate the aircraft that does not know the operation of this product. The pilot will keep the instrument Operating Instructions in the aircraft at all times.

Contents

Warranty	2
1. Important Information and Initial Check Out	3
2. Install the Fuel Flow Transducer	4
3. Install the Fuel Pressure Transducer	6
4. Install the Circular Connector	7
5. Route the Power and Ground Wires	7
6. Route the Backlight Wires	7
7. Route the (Optional) External Warning Control Line	8
8. Route the Fuel Flow Transducer Wires	8
9. (FP-5L Only) Connect the RS-232/422 Input Lines	8
10. (FP-5L Only) Connect the RS-232 Output Line	9
11. Install the Fuel Flow Differential Module (FFDM-I)	9
12. Install the Instrument in the Panel	10
13. Connect the Circular Connector to the Instrument	10
14. System Checkout	10
15. Initial Programming	11
Wiring Diagrams	12-13
Circular Connectors	14-15
Specifications and Operating Features	16-17
Fuel Flow Transducer Installation Drawings	18-21
STC Information	Appendix

Warranty / Agreement

You must read the entire installation and operating instructions. If you do not agree to and accept the terms of this warranty/agreement and the responsibilities set forth in these manuals, DO NOT install this product. Contact E.I. for a refund.

Electronics International Inc. (EI) warrants this instrument and system components to be free from defects in materials and workmanship for a period of one year from the purchase date. EI will repair or replace any item under the terms of this Warranty provided the item is returned to the factory prepaid.

Electronics International Inc. is not liable or responsible for a pilot's action or any situation that results in personal injury, property damage, missed commitments, lack of use of an aircraft or any expenses incurred due to: product failure, inaccuracy in displayed data or text files, display or display format issues, software bugs or problems, upgrade or customization issues, misinterpretation of the display, warning and/or limit settings, calibration problems, installation issues (leaks, mis-wiring, obstructions, damage to aircraft or components, incorrect installation of any parts, wrong parts, parts that don't fit, etc.) or any other issues related to the installation or operation of this product. All of the above are solely the pilot's and/or installer's responsibility. The pilot **must** understand the operation of this product before flying the aircraft. The pilot will not allow anyone to operate the aircraft that does not know the operation of this product. The pilot will keep the instrument Operating Instructions in the aircraft at all times.

By installing this product, the aircraft owner/pilot and installer agree to hold Electronics International Inc. harmless and in no way responsible for monetary compensation, including punitive damages for any incident, harm and/or damage associated with this product (including but not limited to the ones listed above). If you do not agree to the above, **DO NOT INSTALL THIS PRODUCT.**

This Warranty shall not apply to any product that has been repaired or altered by any person other than Electronics International Inc., or that has been subjected to misuse, accident, incorrect wiring, negligence, improper or unprofessional assembly or improper installation by any person. **This warranty does not cover any reimbursement for any person's time for installation, removal, assembly or repair.** Electronics International retains the right to determine the reason or cause for warranty repair.

Personal injury or property damage do to misinterpretation or lack of understanding of this product is solely the pilots responsibility. The pilot **must** understand all aspects of the operation of this product before flying the aircraft. If he/she does not, they agree to seek training from a knowledgeable instructor. Do not allow anyone to operate the aircraft that does not know the operation of this product. **Keep the Operating Instructions in the aircraft at all times.**

This warranty does not extend to any machine, vehicle, boat, aircraft or any other device to which the Electronics International Inc. product may be connected, attached, interconnected or used in conjunction with in any way.

The obligation assumed by Electronics International Inc. under this warranty is limited to repair, replacement or refund of the product, at the sole discretion of Electronics International Inc.

Electronics International Inc. is not liable for expenses incurred by the customer or installer due to factory updates, modifications, improvements, changes, or any other alterations to the product that may affect the form, fit, function or operation of the product.

Electronics International is not responsible for shipping charges or damages incurred under this Warranty.

No representative is authorized to assume any other liability for Electronics International Inc. in connection with the sale of Electronics International Inc. products.

This Warranty is made only to the original user. **THIS WARRANTY IS IN LIEU OF ALL OTHER WARRANTIES OR OBLIGATIONS: EXPRESS OR IMPLIED. MANUFACTURER EXPRESSLY DISCLAIMS ALL IMPLIED WARRANTIES OF MERCHANTABILITY OR FITNESS FOR A PARTICULAR PURPOSE. PURCHASER AGREES THAT IN NO EVENT SHALL MANUFACTURER BE LIABLE FOR SPECIAL, INCIDENTAL OR CONSEQUENTIAL DAMAGES, INCLUDING LOST PROFITS OR LOSS OF USE OR OTHER ECONOMIC LOSS. EXCEPT AS EXPRESSLY PROVIDED HEREIN, MANUFACTURER DISCLAIMS ALL OTHER LIABILITY TO PURCHASER OR ANY OTHER PERSON IN CONNECTION WITH THE USE OR PERFORMANCE OF MANUFACTURER'S PRODUCTS, INCLUDING SPECIFICALLY LIABILITY IN TORT.**

Primary FP-5-LxxHyy and FP-5L-LxxHyy Installation Instructions

I. Important Information and Initial Check Out

- A. The installer and aircraft owner must read the Warranty before starting the installation. There is information in the Warranty that may alter your decision to install this instrument. If you do not accept the terms of the Warranty, do not install this instrument.
- B. If you are not an FAA Certified Aircraft Mechanic familiar with the issues of installing aircraft fuel flow and pressure instruments, Do Not attempt to install this instrument. The installer should use current aircraft standards and practices to install this instrument (refer to AC 43.13).
- C. Check that any necessary FAA Approvals (STCs, etc.) are available for your aircraft before starting the installation. The FAA Approved Model List (AML) is located at the back of this manual. Resolve any issues you may have before starting the installation.
- D. Before starting installation, read the entire Installation Instructions and resolve any installation, operating and performance issues you may have before starting the installation.
- E. **THIS INSTALLATION WILL REQUIRE SOME PARTS UNIQUE TO YOUR AIRCRAFT THAT ARE NOT SUPPLIED IN THE KIT (including, but not limited to hoses and fittings).** Acquire all the parts necessary to install this instrument before starting the installation.
- F. Check that the instrument and flow transducer make and model are correct before starting the installation (check your invoice and the markings on the side of the instrument). **The A gravity feed fuel system or any engine rated over 350 H.P. must use an FT-90 flow transducer. Any engine rated over 550 H.P. must use an FT-180 flow transducer.** A pressure carbureted engine with a fuel return line requires an FFDM-I (see price sheet).

Transducer Identification:

FT-60 - Red Cube
FT-90 - Gold Cube
FT-180 - Black Cube
- G. Before starting the installation make sure the unit will fit in the location you intend to install it without obstructing the operation of any controls.
- H. If this instrument is to replace an existing unit in the aircraft, it is the installer's responsibility to move or replace any existing instruments or components in accordance with FAA approved methods and procedures. The following Installation Instructions do not cover moving or the removal of any existing instruments or components.

I. The fuel pressure limitations for your aircraft are programmed into the unit by the manufacturer. The unit **MUST** have a markings affixed to the case of the instrument which read:

FP-5L-LxxHyy or FP-5-LxxHyy

where "xx" is the low fuel pressure limitation in PSI from the AFM or "00" if no limitations exists and "yy" is the high fuel pressure limitation in PSI from the AFM or "00" if no limitation exists.

J. If the markings described in "I" do not exist, or do not exist in the format shown, or do not match the AFM or POH (as appropriate), do **NOT** install the unit. Contact Electronics International to have the discrepancy resolved.

K. This instrument, when installed in accordance with this STC, will become a primary fuel flow/pressure gauge. If your aircraft has fuel pressure gauged at two locations along the fuel lines, this gauge can be primary for only one of those functions.

L. Check that the limit information on this instrument matches the published limits in your aircraft's P.O.H. or Flight Manual. This information may be listed in the T.C. Data Sheet for your aircraft. Any AD's and/or STC's may set forth additional limitations on the operation of your engine. **It is the aircraft owner's and/or installer's responsibility to determine proper instrument calibration and range markings for your aircraft.**

2. Install the Fuel Flow Transducer

Mount the Fuel Flow Transducer using the appropriate drawing at the back of this manual. **The instructions listed below must be followed when installing a Fuel Flow Transducer.**

<i>Aircraft Configuration</i>	<i>Drawing #</i>	<i>Page</i>
Fuel injected engine without a fuel return line from the fuel servo (most Lycomings).	1229932 or 1229931	18 or 19
Fuel injected engine with a fuel return line from the fuel servo (most Continentals).	0415941	21
Carbureted engine with a fuel pump and no fuel return line.	1229932 or 1229931	18 or 19
Carbureted engine with a fuel pump and a fuel return line (requires an FFDM-1 Module).	1229932 or 1229931, and 1015941	18 or 19, and 20
Carbureted engine with a gravity feed fuel system (requires an FT-90 Flow Transducer).	1229932 or 1229931	18 or 19

Note: If your engine is equipped with a Pressure Carburetor with a fuel return line from the carburetor back to the fuel tank, you will need to install two flow transducers: one in the feed line from the fuel pump to the carburetor and one in the return line from the carburetor back to the fuel tank. Also, a Fuel Flow Differential Module (FFDM-1) will need to be installed. See drawings 1229932 and 1015941 at the back of this manual.

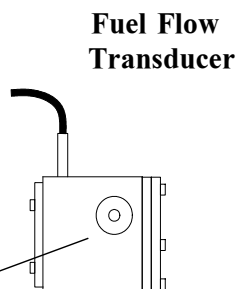
A. The transducer output port should be mounted lower, even or no more than 4" per foot higher than the carburetor inlet port (or fuel servo on a fuel injected engine). If this is not possible, a loop should be put in the fuel line between the Fuel Flow Transducer and the carburetor or fuel servo (see diagram below).

End View

Note: • The direction of the flow of fuel through the transducer is marked on the transducer.

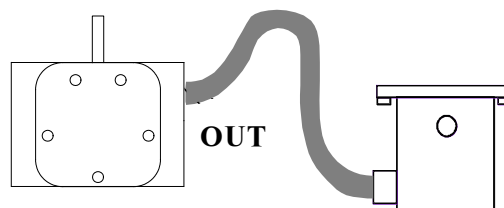
- Mount the transducer with the wires pointing up or the cap with five bolts up or the output up or any combination thereof.

1/4" NPT



Side View

If the transducer is more than 4" per foot higher than the carburetor or fuel servo port, put a loop in the fuel line between the transducer and the carburetor or fuel servo. This will allow bubbles to vacate the Flow Transducer.



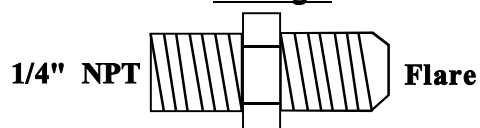
Fuel Flow Transducer

Carburetor or Fuel Servo

- B. Do not remove the yellow caps on the flow transducer until the fuel hoses are ready to be installed.
- C. The flow of fuel through the transducer must follow the direction marked on the transducer.
- D. The flow transducer must be mounted so the wires exiting the transducer are pointing up or the cap with five bolts are pointing up or the output port is pointing up or any combination thereof.
- E. Before connecting any hoses, thoroughly clean them and insure they are free of any loose material. High air pressure may be used, however, do not allow high air pressure to pass through the flow transducer.
- F. When mounting a Fuel Flow Transducer make provisions for the Fuel Pressure Transducer as necessary.

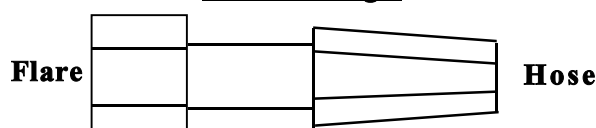
You may want to consider using some Fittings and Hoses shown below. Note: DO NOT EXCEED a torque of 15 ft. lbs. or screw the fittings tighter than two full turns past hand tight, whichever happens first.

Fittings:



- #4 Straight - AN816-4-4 B or C
- #6 Straight - AN816-6 B or C
- #8 Straight - AN816-7 B or C
- #6 45° - MS20823-6 B or C

Hose Fittings:



- Straight - MS24587-XX, Stratoflex 300-, Aeroquip 400-
- 45° - MS27226-XX, Stratoflex 646- and 640, Aeroquip 980006

NOTE: The Stratoflex teflon hose can be much more flexible and easier to route than most existing hoses. If you have a hard to fit installation, consider this hose.

3. Install the Fuel Pressure Transducer

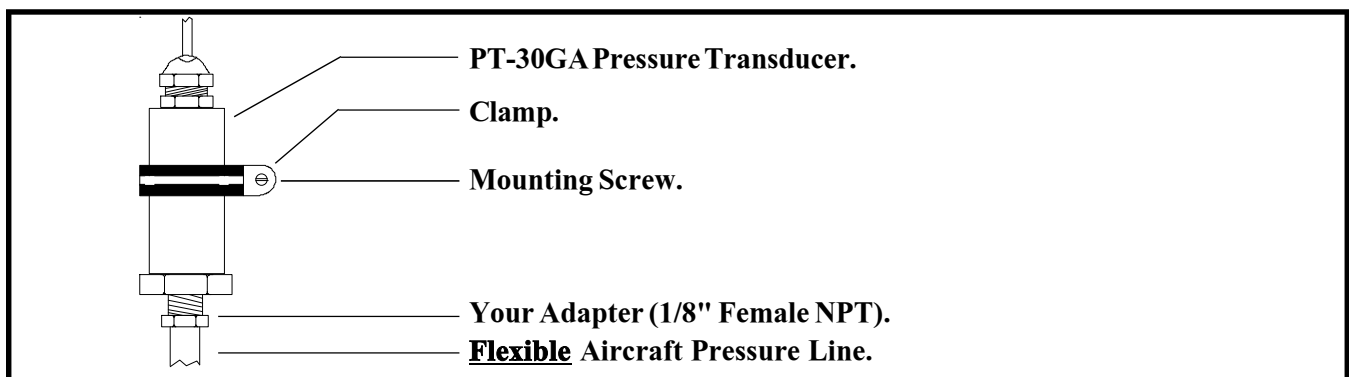
A. Determine the Pressure Pick Up Point

If the current primary fuel flow/pressure indicator to be replaced is measuring metered fuel pressure (pressure at the fuel distributor/spider) and your Pilot Operating Handbook lists fuel pressure limits for this instrument, you must measure fuel pressure at the same pickup point as the original instrument being replaced when installing the PT-30GA Pressure Transducer. Otherwise, you must measure the output of the fuel pump (un-metered pressure) when installing the PT-30GA.

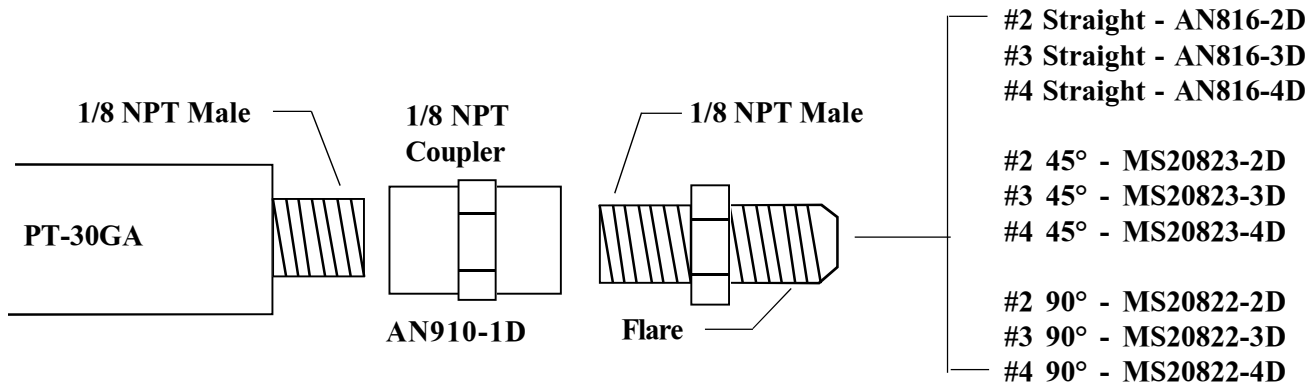
NOTE: Turbocharged aircraft may need to do the following in addition to the instructions above. If your current primary fuel flow/pressure indicator to be replaced measures fuel pressure referenced to the upper deck (or carburetor inlet pressure), you must install a second PT-30GA with EI's Differential Pressure Interface Model (DPIM-1). The second PT-30GA must measure the upper deck or carburetor inlet pressure at the same point as the instrument being replaced. The DPIM-1 allows the FP-5 or FP-5L to read fuel pressure referenced to the upper deck or carburetor inlet pressure. See the supplementary instructions provided with the DPIM-1 for installation instructions.

B. Install the Fuel Pressure Transducer

Find a convenient location on the fire wall and mount the fuel pressure transducer with the clamp provided. **Do not mount the pressure transducer to an engine baffle or directly on the engine supported by an adapter or fitting.** Vibration can cause the adapter to break. The fuel pressure transducer is equipped with a 1/8" NPT male port. This port can be adapted to any fuel pressure line. Use only a flexible hose and fittings suitable for aircraft use. Route a flexible fuel pressure line from the fuel pressure pick up point (as determined in step 3A) to the fuel pressure transducer and tighten all fittings. **Do not use the case of the pressure transducer to tighten the pressure fittings.**



Some Fittings you may want to consider using are shown below:



C. Route the Pressure Transducer Extension Wires

Route the four 6 foot pressure transducer wires (red, black, green and white) in the wire harness to the fuel pressure transducer and cut to length. These wires may be spliced if extra wire length is required. Connect these wires to the pressure transducer with OLC-1 Overlap Connectors, matching the colors of the wires. See OLC-1 Installation Instructions for details. **Tie wrap these wires so they do not obstruct the freedom of travel of any controls.**

4. Install the Circular Connector

Starting from under the instrument panel, route the circular connector wire harness up to the instrument mounting location. (See the Wiring Diagram at the back of this manual). Place the circular connector about 8 inches back from the panel. Tie wrap the harness in place approximately 1 foot back from the circular connector. This will allow the harness to be flexible and accommodate varying lengths in instrument wires. **Be sure these wires do not obstruct the freedom of travel of any controls.**

5. Route the Power and Ground Wires

In the wire harness are two sets of red and black 6' wire bundles used for the fuel pressure transducer and the fuel flow transducer. Also, there are red and black 3' wires used for instrument power and ground. Route the 3' red wire in the harness to the aircraft's 12 or 24 volt main or emergency bus as applicable via an independent circuit breaker (five amps or less). An alternate method would be to route the red lead to the bus via a one amp in-line fuse. **With this method a spare fuse must be kept in the aircraft.** Route the 3' black wire in the harness to a good ground. **Tie wrap these wires so they do not obstruct the freedom of travel of any controls.**

6. Route the Backlight Wires

Connect the backlight wires as follows:

- A. It is recommended to power up the digital display backlight any time the instrument is powered up, although, you can connect the appropriate wires to a panel light rheostat.

1) For a 12-volt system connect the white/brown wire to the bus (or rheostat) and connect the white/red wire to ground (see Wiring Diagram).

2) For a 24-volt system leave the white/brown wire open and connect the white/red wire to the bus (or rheostat) (see Wiring Diagram).

B. Connect the white/orange wire to the panel light rheostat. This wire will dim the Display Mode Indicator LEDs for night operation when the panel lights are turned on. If this line is left open, the Display Mode Indicator LEDs will remain at full intensity at all times. Also, if the voltage on this line drops below 11.5 volts, the analog LEDs will be displayed at full intensity. Tie wrap all wires so they do not obstruct the freedom of travel of any controls.

Note: This line may be connected to the CP-1 Intensity Control Pot (see price sheet).

7. Route the (Optional) External Warning Control Line

The white/yellow wire can be connected to E.I.'s external light (model AL-1), buzzer (model ATG-1), voice annunciator (model AV-17), a relay, etc. This wire grounds when the red warning light is on. The current in this line must be limited to 2/10 of an amp maximum. **Exceeding this limit will damage the instrument.** If this feature is not used, leave this line open. **Tie wrap this wire so it does not obstruct the freedom of travel of any controls.**

8. Route the Fuel Flow Transducer Wires

The wire harness includes 6' red, black and white wires bundled together. Route and connect these 6' wires to the fuel flow transducer using the OLC-1 Overlap Connectors. See OLC-1 Installation Instructions for details. If your engine is equipped with a fuel return line from the carburetor back to the fuel tank, route these wires to the Fuel Flow Differential Module (FFDM-1). See the appropriate drawing at the back of this manual.

Any excess wires can be rolled up and tie wrapped under the instrument panel. **Tie wrap these wires so they do not obstruct the freedom of travel of any controls.** You may decide to cut these wires to a specific length prior to connecting to the fuel flow transducer with the OLC-1 connectors.

9. (FP-5L Only) Connect the RS-232/422 Input Lines

Connecting the FP-5L Input Lines to a compatible GPS unit allows the FP-5L to display Fuel to Destination, Fuel Reserve, Nautical Miles per Gallon and Statute Miles per Gallon information. The FP-5L has three GPS Receive Formats: 1. "In1" for all panel mount GPS units (9600 baud); 2. "In2" for Northstar (1200 baud); 3. "In3" for hand held GPS units (NMEA at 4800 baud). The protocol is 1 start bit, 8 data bits and 1 stop bit and the RS-232 update time of the GPS unit should be 1 to 2 seconds. The GPS unit may require some setup. You may want to contact a knowledgeable instrument shop or the GPS factory to help with the hookup and setup of the GPS unit. See the "Power-Up Programmable Settings" section in the FP-5(L) Operating Instructions to configure the FP-5L RS-232 input.

10. (FP-5L Only) Connect the RS-232 Output Line

Connecting the FP-5L Output Line to a compatible GPS unit allows the GPS unit to use the fuel data transmitted by the FP-5L. The FP-5L has three GPS Transmit Formats: 1. "Ot1" outputs older Shadin fuel flow data (for Arnav, King and newer Garmin GPS units); 2. "Ot2" outputs the Shadin fuel flow sentence (for Garmin and other GPS units); 3. "Ot3" outputs a modified Shadin Fuel/Airdata sentence (for UPS GPS units). The GPS unit may require some setup. You may want to contact a knowledgeable instrument shop or the GPS factory to help with the hookup and setup of the GPS unit. See the "Power-Up Programmable Settings" section in the FP-5(L) Operating Instructions to configure the FP-5L RS-232 output.

Connect the FP-5L RS-232 Output Line (White/Green Wire) to the GPS RS-232 Input Line. Do not connect any GPS shield wires to the FP-5L. They should be left open.

Type of Hook-up	FP-5L Connections	GPS Connections
RS-232	RS-232 Input (white/blue wire)	RS-232 Output
RS-422 or RS-486	RS-232 Input (white/blue wire)	- Output
		+ Output (connect a 120 ohm resistor between the + Output and - Output)
Note: Do not connect any GPS shield wires to the FP-5L. They should be left open.		

11. Install the Fuel Flow Differential Module (FFDM-1)

If your engine is equipped with a fuel return line from the carburetor back to the fuel tank, install the FFDM-1 in the aircraft as outlined below (see diagram at the back of this manual). Otherwise, omit this step.

- A. Connect the circular connector to the FFDM-1.
- B. Install the FFDM-1 under the instrument panel using two tie wraps on each end of the module to support it to a wire bundle or bracket.
- C. Route and connect the 3' red power lead to the 12 or 24 volt bus via a 1 amp fuse.
- D. Route and connect the 3' black ground lead to the same ground used for the FP-5.
- E. Route and connect the 6' red, black and white leads marked "Feed" to the flow transducer installed in the fuel line from the fuel pump to the carburetor, using OLC-1 Overlap Connectors supplied with the transducer. See OLC-1 Installation Instructions for details.
- F. Route and connect the 6' red, black and white leads marked "Return" to the flow transducer installed in the return fuel line from the carburetor to the fuel tank, using OLC-1 Overlap Connectors supplied with the transducer. See OLC-1 Installation Instructions for details.
- G. Connect the 1' red, black and white leads to the same color 6' leads from the FP-5.

H. Any excess wires can be rolled up and tie wrapped under the instrument panel. **Tie wrap these wires so they do not obstruct the freedom of travel of any controls.** You may decide to cut these wires to a specific length prior to connecting to the fuel flow transducer with the OLC-1 connectors.

Note: The flow transducers for the FFDM-1 and the FP-5 **MUST** be of the same model (i.e., if the FP-5 uses an FT-60 flow transducer, then the FFDM-1 must use a FT-60 flow transducer).

12. Install the Instrument in the Panel

- A. Install the instrument from behind the instrument panel using 6 x 32 screws. **These screws must not be any longer than 1/2".** Tie wrap any loose wires as needed. Make sure the instrument and wire do not obstruct the operation of any controls.
- B. Mount the placard "Do Not Rely on Fuel Flow Instrument to Determine Fuel Levels in Tanks" on the aircraft instrument panel near the FP-5.
- C. If the FP-5(L) is replacing a fuel flow gauge which displays in fuel flow and the Pilot Operating Handbook lists **fuel flow limits**, you **must** create a placard indicating those flow limits. The placard must be mounted near the FP-5(L).

13. Connect the Circular Connector to the Instrument

- A. Push the two mating connectors together and twist them until they snap into position.
- B. Turn the locking ring on the instrument connector clockwise (1 1/2 turns) until it locks into position.

14. System Checkout

Check instrument operation as follows:

- A. Turn the aircraft master switch on (engine off) and verify that the red warning LEDs on the FP-5 flash and the green "REM" mode LED is blinking. A problem at this step could be caused by poor connections on the red or black power and ground leads.
- B. Set the instrument toggle switch to "FLOW" and check for a digital fuel flow reading of "000." A problem at this step could be caused by a poor connection or crossed flow transducer wires. The voltage on the flow transducer wires (with the transducer removed from the instrument) should measure as follows:

Red Wire	-	+9 to 14 Volts
Black Wire	-	0 Volts
White Wire	-	0 or 5 Volts (pulsed when fuel is flowing)

- C. Check the digital display backlight. With high or medium ambient light it is hard to see the digital display backlight (it is only required during low ambient light conditions but should be on at all times).

- D. If the Display Mode Indicator LED dimming wire has been connected, turn the panel light rheostat up and look for the Display Mode Indicator LEDs to dim.**
- E. With the engine running, check the "FLOW" Display Mode to read properly. If there is a problem at this point see step B above for troubleshooting information. To see if the instrument is receiving pulses from the flow transducer, disconnect the white wire from the transducer and short it rapidly (white wire to the instrument) to ground. A reading should appear on the display.**
- F. (FP-5L Only) Check the FP-5L display to read a number when the "F. to D." (Fuel to Destination) button is pushed. You may have to fly the aircraft before the GPS unit will output data. If the "F. to D." function is not working properly, use the following chart to help find your problem.**

FP-5L Display	Comments
Off	The FP-5L is not receiving serial data. Check Connections and the setup of the Loran/GPS unit.
' on (note the bar)	The FP-5L is receiving serial data but it does not have the proper protocol. Check connections the Loran/GPS Interface settings on the FP-5L.
on	The FP-5L is receiving RS-232 data but the Speed and/or Distance data is missing. Check the setup of the Loran/GPS unit.

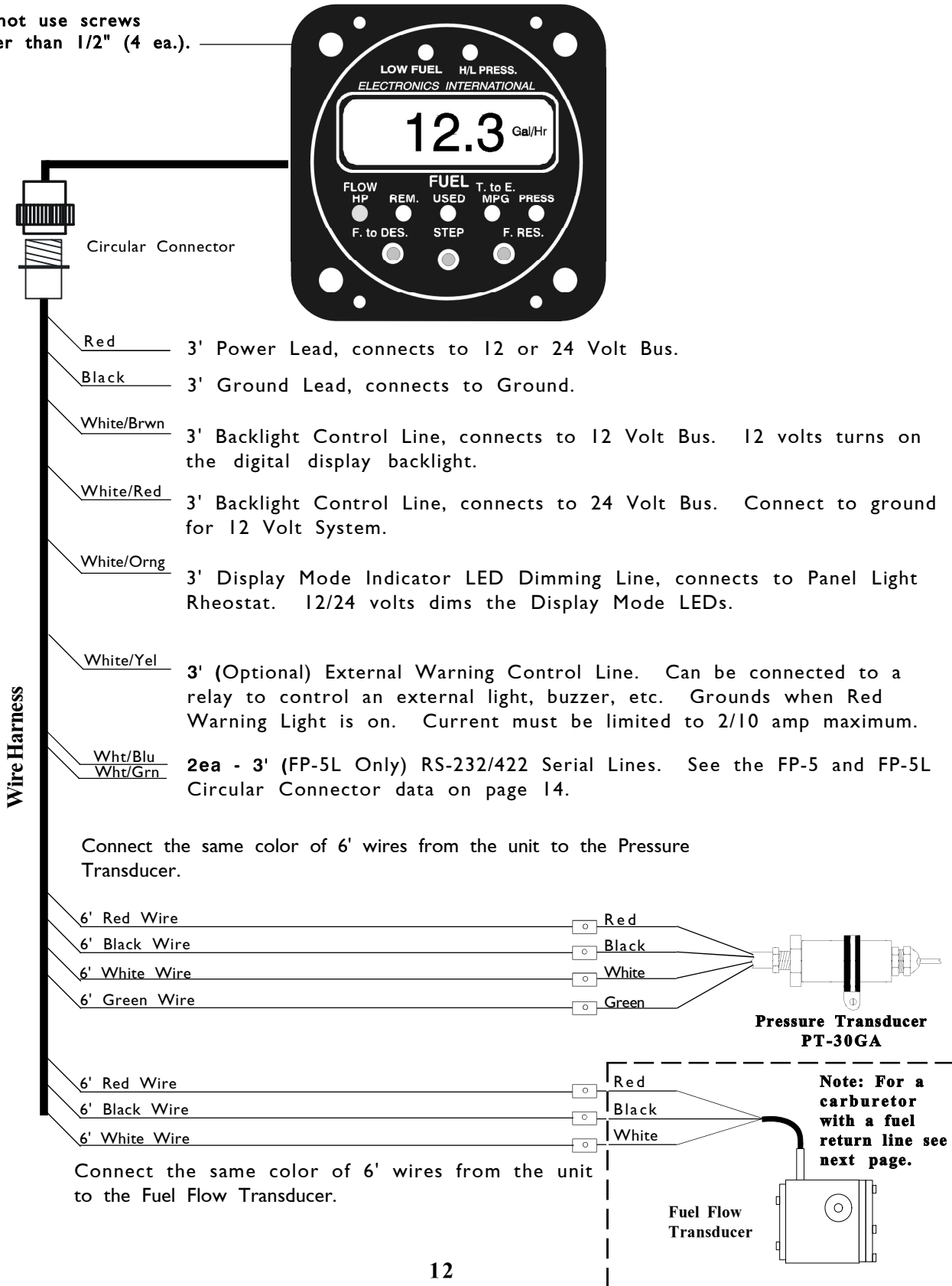
- G. After running the engine, check the fuel hoses, transducers and fittings for leaks.**

15. Initial Programming

The Power-Up Programmable Settings for the FP-5(L) must be set up for your aircraft. See the Power-Up Programmable Setting section in the Operating Instruction manual for set up information.

Fuel Flow/Pressure (FP-5 and FP-5L) Wiring Diagram

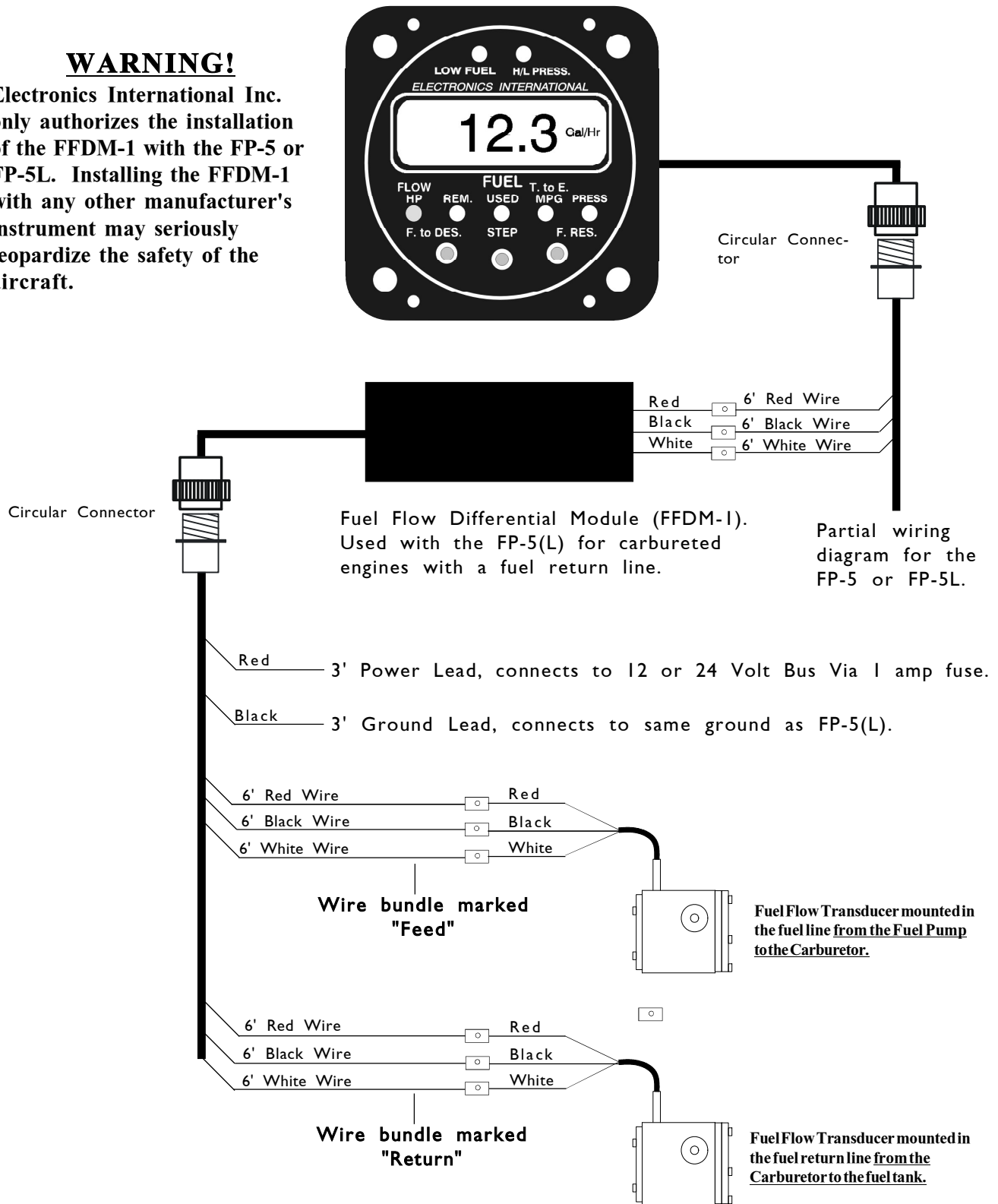
Do not use screws
longer than 1/2" (4 ea.).



FP-5(L) / FFDM-1 Interconnect Wiring Diagram

WARNING!

Electronics International Inc. only authorizes the installation of the FFDM-1 with the FP-5 or FP-5L. Installing the FFDM-1 with any other manufacturer's instrument may seriously jeopardize the safety of the aircraft.



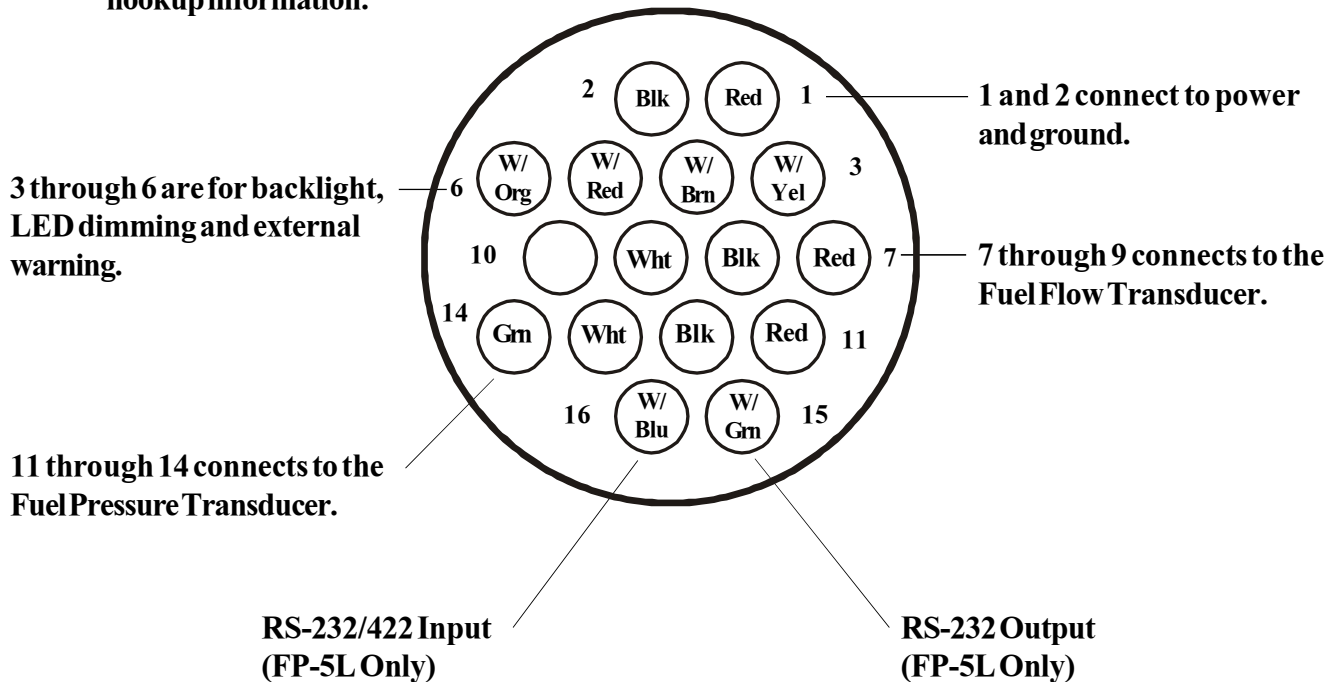
FP-5 and FP-5L Circular Connector

Connecting Cable Harness, Back View (wire side)

OR

Instrument Connector, Front View

Note: See Wiring Diagram for hookup information.



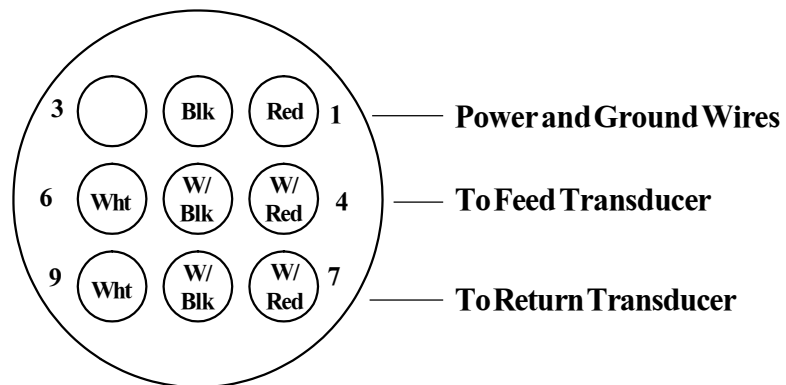
FFDM-1 Circular Connector

Connecting Cable Harness, Back View (wire side)

OR

Module Connector, Front View

**Note: See Wiring Diagram
for hook up information.**



Specifications and Operating Features

Model

FP-5 and FP-5L (Fuel Flow/Pressure Instrument)

Case Dimensions

2.5" x 2.5" x 3.65" depth, 2 1/4" Bezel.

Weight

Instrument Only:	11 Oz.
Flow Transducer FT-60, FT-90 or FT-180:	6 Oz.
Fuel Pressure Transducer PT-30GA:	2 Oz.

Environmental

Meets TSO C44a/C47

Power Requirements

7.5 to 35 Volts, 1/10 Amp.

Green Display Mode Indicator LEDs

The intensity of these LEDs is controlled by the dimming wire. 12 or 24 volts on this wire will dim the LEDs for night operation.

Red Low Fuel Warning LED

This LED will blink any time the programmed First or Second Low Fuel limit, Time to Empty Limit or Reoccurring Alarm is violated. The Low Fuel Warning LED is always displayed at full intensity and will flash on power-up.

Red H/L PRESS Warning LED

This LED will blink any time the factory programmed High or Low Pressure limit is violated. The H/L PRESS Warning LED is always displayed at full intensity and will flash on power-up.

Digital Display

LCD (viewable in direct sunlight), with 12 and 24 volt backlight control wires for night operation. Displays "8888" on power up.

External Warning Control Line

Grounds when any Red Warning LED is on or blinking. Current should be limited to 2/10 amp.

Accuracy

Flow:	2% or better in accordance with TSO C44a.
Pressure:	2% in accordance with TSO.

Resolution

Fuel Flow:	0.1 Gal. or 1 Lb. or 1 Ltr.
Fuel Remaining:	0.1 Gal. up to 99.9 Gal or 1 Lb. or 1 Ltr.
Fuel Used:	0.1 Gal. up to 99.9 Gal or 1 Lb. or 1 Ltr.
Time to Empty:	1 minute
Fuel Pressure:	0.2 PSI (with PT-30GA)

Max Displayed Range (Unit Only)

Fuel Flow: 199.9 Gals/Hr or 162.0 br Gal/Hr or 1199 Lbs/Hr or 749 Ltr/Hr
Fuel Remaining: 999 Gals. or 811 br Gal. or 1999 Lbs. or 1999 Ltr.
Fuel Used: 999 Gals. or 811 br Gal. or 1999 Lbs. or 1999 Ltr.
Time to Empty: 19 hours 59 minutes
Fuel Pressure: 0.5 to 199.8 PSI (Readings below 0.5 PSI will be displayed as 00.0 PSI)

RS-232/422 Input Ports (FP-5L Only)

Single Line Receive Method: RS-232C or RS-423
Dual Line Receive Method: RS-422 or RS-485 (with 120 ohm external resistor)
Protocol: 1 Start bit, 8 Data bits, 1 Stop bit
Baud Rate: 1200, 4800, 9600
Receive Format: Moving Map, Northstar or NMEA

RS-232/422 Output Port (FP-5L Only)

Transmit Method: RS-232C Single Line
Protocol: 1 Start bit, 8 Data bits, 1 Stop bit
Baud Rate: 9600 (Receive Format must be set to Moving Map)
Transmit Format: King KLN88, Garmin, or UPS

Fuel Flow Transducer, FT-60 (Red Cube)

Range: 0.6 to 70+ GPH
Linearity: 1% over an engine's normal operating range
K Factor: Approx. 68,000
Pressure Drop: 0.5 PSI at 28 GPH
2.0 PSI at 56 GPH
Working Press: 1000 PSI
Min. Burst Press: 4000 PSI
Temp. Range: -65°C to 125°C
Fuel Ports: 1/4" Female NPT

Fuel Flow Transducer, FT-90 (Gold Cube)

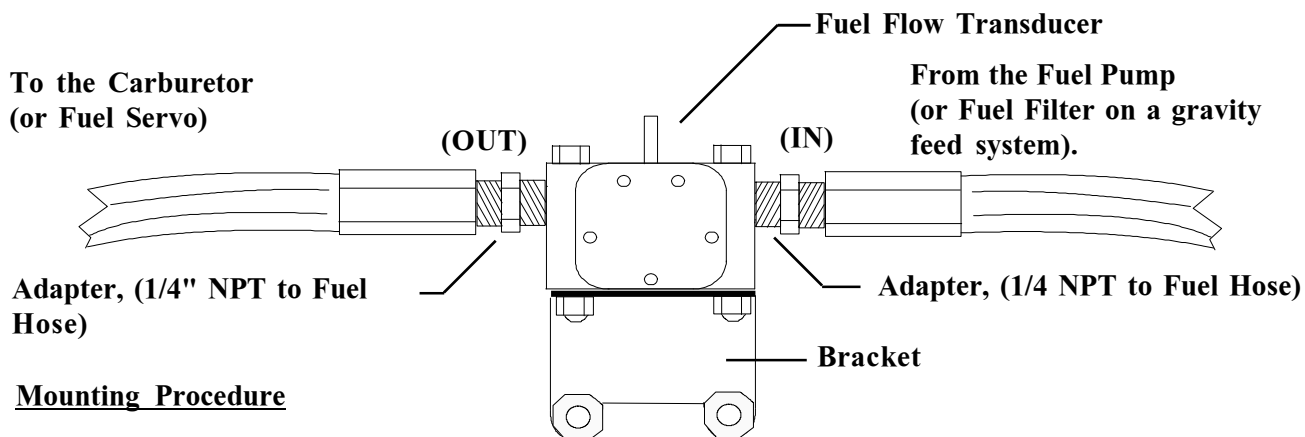
Range: 2 to 125+ GPH
K Factor: Approx. 33,800
Pressure Drop: 0.5 PSI at 63 GPH
2.0 PSI at 127 GPH
Working Press: 1000 PSI
Min. Burst Press: 4000 PSI
Temp. Range: -65°C to 125°C
Fuel Ports: 1/4" Female NPT

Fuel Flow Transducer, FT-180 (Black Cube)

Range: 2 to 250 GPH
K Factor: Approx. 22,400
Pressure Drop: 0.5 PSI at 88 GPH
2.0 PSI at 176 GPH
Working Press: 1000 PSI
Min. Burst Press: 4000 PSI
Temp. Range: -65°C to 125°C
Fuel Ports: 1/4" Female NPT with #8
Female Flare Fitting

Fuel Pressure Transducer (PT-30GA)

Range: 0 to 40 PSI
Over Press: 90 PSI without damage.
Min. Burst Press: 120 PSI
Temp. Range: -40°C to 125°C
Material: 303 Stainless Steel
Press. Port: 1/4" Male NPT



Mounting Procedure

1. Find a convenient location on the fire wall (away from any hot exhaust pipes) and mount a bracket for the Fuel Flow Transducer. Check both sides of the fire wall for clearance before drilling any holes.
2. Mount the Fuel Flow Transducer onto the Bracket. **You must use the FT-90 (Gold Cube) Fuel Flow Transducer on a gravity feed system or for any engine over 350 H.P.** If the Transducer is mounted within 6" of an exhaust pipe, the Flow Transducer must be wrapped with Fire Sleeving.
3. Remove the fuel hose which goes from the Fuel Pump (or the Fuel Filter on a gravity feed system) to the Carburetor (or Fuel Servo).
4. Purchase two new hoses, one from the fuel pump (or the Fuel Filter) to the Fuel Flow Transducer (making provisions for the fuel pressure transducer as necessary) and the other from the Fuel Flow Transducer to the carburetor (or Fuel Servo). **There must be flexible hose in and out of the Transducer.** The hoses must meet TSO-C53a Type C or D FAA specification. **The new hoses must be the same size as the current hoses in the aircraft.** A source of fittings and fabricated hoses is:

Sacramento Sky Ranch Inc.
(916) 421-7672
(800) 433-3564
Fax: (916) 421-5719

OR

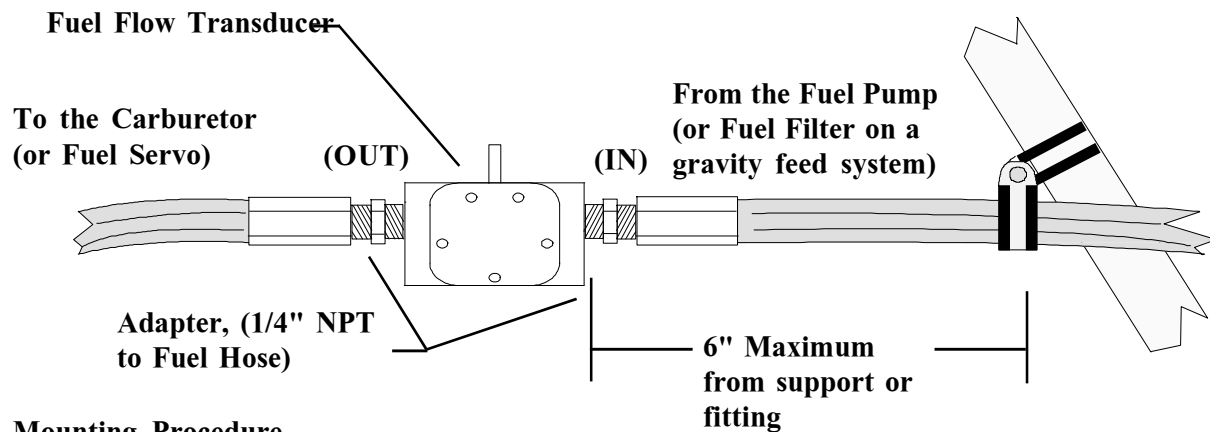
Varga Enterprises Inc.
(480) 963-6936
(800) 966-6936
FAX: (480) 899-0324

OR

Hoses Unlimited Inc.
(510) 483-8521
Fax: (510) 483-8524

5. **Read the Installation Instructions for important installation considerations.**

Drawn By: R.R.	<i>Electronics International Inc.</i>		
Approved By: R.R.	Installation of a Fuel Flow Transducer on the Fire Wall and in the fuel line <u>from the fuel pump to the carburetor or fuel servo.</u> Note: <u>Not applicable</u> for a fuel injected engine with a fuel return line (see D/N 0415941).		
Scale: None			
Material:			
Next Assembly:			
P/N:	Date: 12/29/93	Rev: D: 7/2/02	D/N: 1229931



Mounting Procedure

1. Find a convenient location within 6" of a hose support or fitting and away from any hot exhaust pipes to suspend the Fuel Flow Transducer. The hose support or fitting may be on the input or output line of the Flow Transducer.
2. Remove the fuel hose which goes from the Fuel Pump (or the Fuel Filter on a gravity feed system) to the Carburetor (or Fuel Servo).
3. Purchase two new hoses: one to be used from the fuel pump (or the Fuel Filter) to the Fuel Flow Transducer and the other to be used from the Fuel Flow Transducer to the carburetor (or Fuel Servo). **There must be flexible hose in and out of the Transducer.** The hoses must meet TSO-C53a Type C or D FAA specification. **The new hoses must be the same size as the current hoses in the aircraft.** A source of fittings and fabricated hoses is:

Sacramento Sky Ranch Inc.

(916) 421-7672

(800) 433-3564

Fax: (916) 421-5719

OR

Varga Enterprises Inc.

(480) 963-6936

(800) 966-6936

FAX: (480) 899-0324

OR

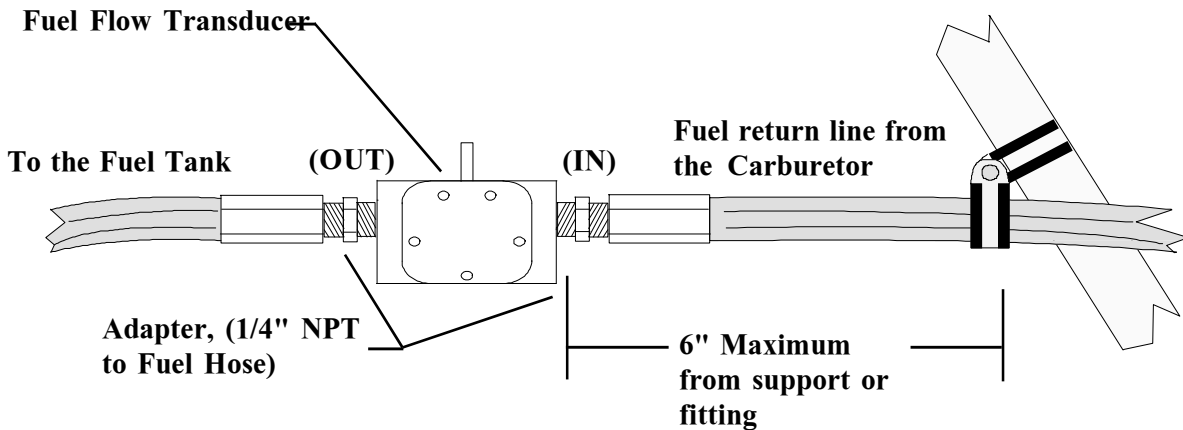
Hoses Unlimited Inc.

(510) 483-8521

Fax: (510) 483-8524

4. Mount the Fuel Flow Transducer in the fuel line. **You must use the FT-90 (Gold Cube) Fuel Flow Transducer on a gravity feed system or for any engine over 350 H.P. You must use the FT-180 (Black Cube) for any engine over 550 H.P.** If the Transducer is mounted within 6" of an exhaust pipe, the Flow Transducer must be wrapped with Fire Sleeving.
5. **Read the Installation Instructions for important installation considerations.**

Drawn By: R.R.	<i>Electronics International Inc.</i>		
Approved By: R.R.	Installation of a Fuel Flow Transducer suspended in the fuel line <u>from the fuel pump to the carburetor or fuel servo.</u> Note: <u>Not applicable</u> for a fuel injected engine with a fuel return line (see D/N 0415941).		
Scale: None			
Material:			
Next Assembly:			
P/N:	Date: 12/29/93	Rev: D: 7/2/02	D/N: 1229932



Mounting Procedure

1. Find a convenient location within 6" of a hose support or fitting and away from any hot exhaust pipes to suspend the Fuel Flow Transducer. The hose support or fitting may be on the input or output line of the Flow Transducer.
2. Remove the return fuel hose which goes from the Carburetor to the Fuel Tank.
3. Purchase two new hoses: one to be used from the Carburetor to the Fuel Flow Transducer and the other to be used from the Fuel Flow Transducer to the Fuel Tank. **There must be flexible hose in and out of the Transducer.** The hoses must meet TSO-C53a Type C or D FAA specification. **The new hoses must be the same size as the current hose in the aircraft.** A source of fittings and fabricated hoses is:

Sacramento Sky Ranch Inc.
 (916) 421-7672
 (800) 433-3564
 Fax: (916) 421-5719

Varga Enterprises Inc.
 OR (480) 963-6936
 (800) 966-6936
 FAX: (480) 899-0324

Hoses Unlimited Inc.
 (510) 483-8521
 Fax: (510) 483-8524

4. Mount the Fuel Flow Transducer in the fuel return line. **You must use the FT-90 (Gold Cube Fuel Flow Transducer on any engine that has over 350 H.P. You must use the FT-180 (Black Cube) for any engine over 550 H.P.** If the Transducer is mounted within 6" of an exhaust pipe, the Flow Transducer must be wrapped with Fire Sleeving.
5. **Read the Installation Instructions for important installation considerations.**

Drawn By: R.R.	<i>Electronics International Inc.</i>		
Approved By: R.R.	Installation of a Fuel Flow Transducer suspended in the <u>fuel return line</u> from the carburetor to the fuel tank. Note: <u>Only applicable</u> for installation on aircraft with a fuel return line from the <u>Carburetor</u> .		
Scale: None			
Material:			
Next Assembly:			
P/N:	Date: 10/15/94	Rev: A: 7/2/02	D/N: 1015941

Mounting Procedure

1. Find a convenient location between the Fuel Servo and Flow Divider and away from any hot exhaust pipes to suspend the Fuel Flow Transducer.
2. Remove the fuel hose which goes from the Fuel Servo to the Flow Divider.
3. Purchase two new hoses: one to be used from the Fuel Servo to the Fuel Flow Transducer and the other to be used from the Fuel Flow Transducer to the Flow Divider. **There must be flexible hose in and out of the Fuel Transducer.** The hoses must meet TSO-C53a Type C or D FAA specification. **The new hoses must be the same size as the current hose in the aircraft.** A source of fittings and fabricated hoses is:

Sacramento Sky Ranch Inc.
(916) 421-7672
(800) 433-3564
Fax: (916) 421-5719

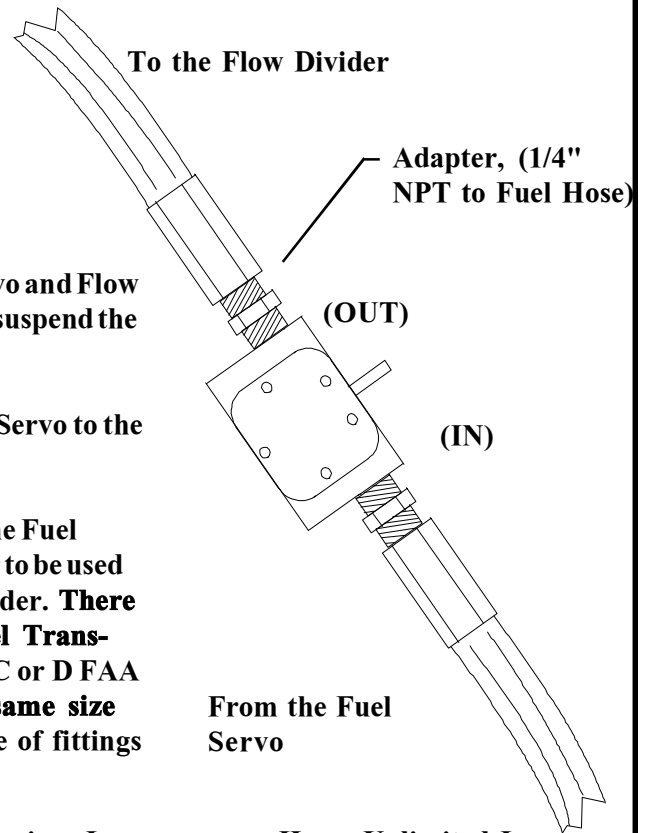
OR

Varga Enterprises Inc.
(480) 963-6936
(800) 966-6936
FAX: (480) 899-0324

OR

Hoses Unlimited Inc.
(510) 483-8521
Fax: (510) 483-8524

4. Mount the Fuel Flow Transducer in the fuel line. **You must use the FT-90 (Gold Cube) Fuel Flow Transducer on any engine over 350 H.P. You must use the FT-180 (Black Cube) for any engine over 550 H.P.** If the Transducer is mounted within 6" of an exhaust pipe, the Flow Transducer must be wrapped with Fire Sleeving.
5. **Read the Installation Instructions for important installation considerations.**



Drawn By: R. R.	<i>Electronics International Inc.</i>		
Approved By: R.R.	Installation of the Fuel Flow Transducer suspended in the fuel line <u>between the Fuel Servo and the Flow Divider.</u>		
Scale: None			
Material:	Note: <u>Only applicable</u> for installation on aircraft with a fuel return line from the Fuel Servo.		
Next Assembly:			
P/N:	Date: 4/15/94	Rev: B: 7/2/02	D/N: 0415941

United States of America
Department of Transportation Federal Aviation Administration
Supplemental Type Certificate

Number SA01157LA

This certificate, issued to **Electronics International, Inc.
63296 Powell Butte Hwy.
Bend, OR 97701**

*certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part * of the * Regulations.*

Original Product—Type Certificate Number: * See attached FAA Approved Model List (AML)
Make: No. SA01157LA for a list of approved aircraft
Model: models and applicable airworthiness TCDS

Description of the Type Design Change: Installation of Electronics International Model FP-5 or FP-5L Fuel Flow/Fuel Pressure Instrument as a replacement instrument in accordance with document II SO506931 revisions listed on the attached AML No. SA01157LA, or later FAA approved revision.

Limitations and Conditions: The installation should not be incorporated in any aircraft unless it is determined that the interrelationship between this installation and any previously approved configuration will not introduce any adverse effect upon the airworthiness of the aircraft. The approval of this modification applies to the above noted airplane model series only. A copy of this STC, the AML, and Airplane Flight Manual Supplement, AFM2112, Rev. B, or later FAA approved revision must be included in the permanent records of the modified aircraft. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: September 11, 2000

Date reissued: December 3, 2001, January 20, 2005

Date of issuance: June 1, 2001

Date amended: January 20, 2005



By direction of the Administrator

(Signature)

Acting Manager, Seattle Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both

This certificate may be transferred in accordance with FAR 21.47.

Department of Transportation—Federal Aviation Administration

Supplemental Type Certificate

(Continuation Sheet)

Number SA01157LA

Electronics International, Inc.

Reissued: December 3, 2001, January 20, 2005

Amended: January 20, 2005

Limitations and Conditions: (cont'd)

In locations where this STC removes an existing combination manifold pressure (MAP)/fuel flow instrument, an FAA approved MAP gage must be installed in the aircraft with the FP-5 or FP-5L fuel flow/fuel pressure instrument installed by this STC.

This instrument must not be the primary source of information for determining the fuel quantity on board.

The instrument fuel pressure limits must be programmed at the factory and may not be pilot programmable.

- END -

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA FORM 8110-2-1 (10-69)

This certificate may be transferred in accordance with FAR 21.47.

FAA Approved Model List (AML) SA01157/LA

For Installation of

Electronics International, Inc. Primary Fuel Flow/Pressure Instruments

Issue Date: June 1, 2001

Item	Aircraft Make	Aircraft Model	Original Type Certificate Number	Certification Basis For Alteration	Flight Manual Supplement		Installation Instructions		AML Amended Date
					Number	Revision	Number	Revision	
					AFM2112	B 1/07/2005 or Later FAA Approved Revision	II S0506931	B 709/2004 or Later FAA Approved Revision	
1	Aero Commander (Volare)	10, 10A, 100, 100A, 100-180	1A21	CAR 3					1/20/2005
2	Aeromot	AMT-100, AMT-200, AMT-200S, AMT-300	TG00004AT	CPR 21					
3	Aerospatiale	See Social							
4	Alliance Aircraft Group (Helio Enterprises)	H-250, H-295, HT-295	1A8	CAR 3					
		H-391B, H-395, HB95A, H-391, H-700, H-800	1A8	CAR 3					
5	Alon	See Unisair							
6	American Blimp	A-60, A-60+	ASIRNM	FAR 21					
		A-1-50	S00002SE	FAR 21					
7	American Champion (Aeronca, Bellanca, Trytek)	7BCM, 7DC, S7DC	A-759	CAR 4A					
		7CCM, S7CCM	A-759	CAR 4A					
		7EC, 7ECA, S7EC, 7HC, 7KC, 7KCAB	A-759	CAR 4A					
		7GC, 7GCA, 7GCAA, 7GCB, 7GCEA, 7GCBC	A-759	CAR 4A					
		7GCBA	A-759	CAR 8					
		8KCAB, 8GCBC	A2ICE	FAR 23					
		11AC, 11BC, S11AC, S11BC	A-761	CAR 4A					
		11CC, S11CC	A-796	CAR 3					
8	American General Aircraft	See Gulfstream American							
9	Augustair Inc. (Varga)	2180	4A19	CAR 3					
10	Aviat Inc. (Pitts, Sky, Child F. Doyle)	A-1, A-1A, A-1B	A22NM	FAR 23					
		S-1S, S-1T	AS80	FAR 23					
		S-2A, S-2S, S-2B, S-2C	AS80	FAR 23					
11	Beech Aircraft Corp.	18A, S18A	639	BUL 7A					
		19A, 23, A33, A23A-A23, C23, B23, A33-24	A1CE	CAR 3					
		A24, A24R, B24R, A23-19	A1CE	CAR 3					
		35, 35R, A35, B35, C35, D35, E35, F35, G35	A-771	CAR 3					
		HB35, JB35, KB35, MB35, NB35, PB35, VB35A, VB35B, 35-33	3A15	CAR 3					
		35-A33, 35-B33, 35-C33, 35-C33A, E33, E33A, E33C, F33	3A15	CAR 3					
		FE3A, FE3C, G33, 36, A36, A36TC, B36TC	3A15	CAR 3					
		50, B50, C50, D50, D50A, D50B, D50C	5A4	CAR 3					
		D50E, D50E-5990, E50, F50, G50, H50, J50	5A4	CAR 3					
		D55, D55A, E55, E55A	3A16	CAR 3					
		56TC, A56TC	3A16	CAR 3					
		58, 58A	3A16	CAR 3					
		58P, 58PA, 58TC, 58TCA	A23CE	FAR 23					
		60, A60, B60	A12CE	FAR 23					
		65, 65-80, 65-A80, 65-85, 65-880, A65, A65-8200, 70	3A20	CAR 3					
		76	A29CE	FAR 23					
		77	A30CE	FAR 23					

FAA Approved Model List (AML) SA01157LA For Installation of Electronics International, Inc. Primary Fuel Flow/Pressure Instruments Issue Date: June 1, 2001									
Item	Aircraft Make	Aircraft Model	Original Type Certificate Number	Certification Basis For Alteration	Flight Manual Supplement		Installation Instructions		AML Amended Date
					Number	Revision	Number	Revision	
					AFM2112	B 1072005 or Later FAA Approved Revision	II S0506931	B 70692004 or Approved Revision	
12	Bellanca Aircraft Corporation (See American Champion)	95, B95, B95A, D95A, E95, 95-55, 95-A55, 95-B55	3A16	CAR 3	"	"	"	"	"
		95-B55A, 95-B55B, 95-C55, 95-C55A	3A16	CAR 3	"	"	"	"	"
		17-30A, 17-31A	A18CE	FAR 23	"	"	"	"	"
		17-31ATC	A18CE	FAR 23	"	"	"	"	"
		14-19, 14-19-2, 14-19-3, 14-19-3A, 17-30, 17-31, 17-31TC	1A3	CAR 3	"	"	"	"	"
13	Boeing Aircraft	75 thru E75, A75J1, A75L300, A75SN1 thru E75N1, JB75A	A-745	CAR 4A	"	"	"	"	"
14	Cessna Aircraft Corp.	170 thru 170B	A-799	CAR 3	"	"	"	"	"
		172 thru 172Q	3A12	CAR 3	"	"	"	"	"
		172R, 172S	3A12	FAR 23	"	"	"	"	"
		R172E thru R172K, 172RG	3A17	CAR 3	"	"	"	"	"
		175 thru 175C	3A17	CAR 3	"	"	"	"	"
		177A, 177B	A13CE	FAR 23	"	"	"	"	"
		177RG	A20CE	FAR 23	"	"	"	"	"
		TR182	3A13	CAR 3	"	"	"	"	"
		185, 185A, 185B	3A24	CAR 3	"	"	"	"	"
		185C, 185D, 185E, A185E, A185F	3A24	CAR 3	"	"	"	"	"
		188, 188A, 188B, A188, A188A, A188B, T188C	A9CE	FAR 23	"	"	"	"	"
		206, 206H, P206 thru P206E, T206H	A4CE	CAR 3	"	"	"	"	"
		TP206A thru TP206E	A4CE	CAR 3	"	"	"	"	"
		U206 thru U206G, TU206A thru TU206G	A4CE	CAR 3	"	"	"	"	"
		207, 207A, T207, T207A	A15CE	FAR 23	"	"	"	"	"
		210, 210A, 210B, 210C, 210D, 210E, 210F, 210G, T210G	3A21	CAR 3	"	"	"	"	"
		210H, T210H, 210I, T210I, 210K, T210K, 210L, T210L	3A21	CAR 3	"	"	"	"	"
		210M, T210M, 210N, T210N, 210P, T210P, 210R	3A21	CAR 3	"	"	"	"	"
		210-5 (205), 210-5A (205A)	3A21	CAR 3	"	"	"	"	"
		T303	A34CE	FAR 23	"	"	"	"	"
		305A, 305C, 305D, 305F	5A5	CAR 3	"	"	"	"	"
		305B, 305E	3A14	CAR 3	"	"	"	"	"
		310, 310A, 310B, 310C, 310D, 310E, 310F, 310G, 310H, E310H	3A10	CAR 3	"	"	"	"	"
		310L, 310I, 310J-1, E310I, 310K, 310L	3A10	CAR 3	"	"	"	"	"
		310N, 310P, 310R, T310P, T310R, 310Q, T310Q	3A10	CAR 3	"	"	"	"	"
		320, 320-1, 320A thru 320F, 335, 340, 340A	3A25	CAR 3	"	"	"	"	"
		336, 337, 337A, 337B, 337C, 337D, 337E, 337F, 337G, 337H	A6CE	CAR 3	"	"	"	"	"
		M337B, P337H	A6CE	CAR 3	"	"	"	"	"
		T337B, T337C, T337D, T337E, T337F, T337G, T337H	A6CE	CAR 3	"	"	"	"	"
		T337H-SP, 340, 340A	A6CE	CAR 3	"	"	"	"	"
		401, 401A, 401B, 402, 402A, 402B, 402C	A7CE	CAR 3	"	"	"	"	"

FAA Approved Model List (AML) SA01157LA

For Installation of Electronics International, Inc. Primary Fuel Flow/Pressure Instruments

Issue Date: June 1, 2001

Item	Aircraft Make	Aircraft Model	Original Type Certificate Number	Certification Basis For Alteration	Flight Manual Supplement Number	Revision	Installation Instructions Number	Revision	AML Amended Date
15	Child Doyle F.	See <i>Amair</i>							
16	Christen Industries	See <i>Amair</i>							
17	Cirrus Design Corporation	SR20, SR22	A00009CH	FAR 23					10/17/2005
18	Clark	1000	2A6	CAR 8					1/20/2005
		12	2A12	CAR 8					
19	Commander Aircraft	112, 112B, 114, 114A, 112TC, 112TCA	A12S0	FAR 23					6/1/2001
		114B	A12S0	FAR 23					1/20/2005
		500, 500-A, 500-B, 500-S, 500-U, 520, 560, 560-A, 560-B	6A1	CAR 3					
		560-F, 680, 680-E, 680-F, 680-FL, 680-FL(P), 685	2A4	CAR 3					
		700	A12SW	FAR 23					
		720	2A4	CAR 3					
20	DeHavilland Aircraft	DH 82A	A5PC	FAR 21					1/10/2006
		DH 82A	A8EU	FAR 21					
		DH C1 Chipmunk 21, DH C1 Chipmunk 22A	A44EU	FAR 21					
		DH C1 Chipmunk 22	A44EU	FAR 21					
		DHC-1B-2-S3, DHC-1B-2-S5	A26NM	FAR 21					
		DHC-2 Mk. I (L-20A), DHC-3 Mk II	A-806	CAR 10					
		DHC-3	A-815	CAR 10					
		L-20A	AR-33	CAR 8					
21	Diamond Aircraft Industries	DA 20-A1, DA 20-C1	TA-CH	FAR 21					1/20/2005
		DA 40	A47CE	FAR 21					10/17/2005
22	Dornier-Werke	DO 27 Q-6	A8IN	CAR 10					1/20/2005
		DO 28 A-1, DO 28 B-1	7A13	CAR 10					
		DO 28 D, DO 28 D-1	A16EU	FAR 23					
		228-100, 228-101, 228-200, 228-201, 228-202, 228-212	A16EU	FAR 23					
23	ERCO	See <i>Unair</i>							
24	Extra Flugzeugbau	EA-200, EA-300, EA-300L	A67EU	FAR 23					
		EA-400	A43CE	FAR 21					10/17/2005
25	Forney	See <i>Unair</i>							
26	Globe (Swift)	GC-1A, GC-1B	A-766	CAR 4A					1/20/2005
27	Gulfstream American Corp.	AA-1, AA-1A, AA-1B, AA-1C	A11EA	FAR 23					10/17/2005
	(American General, Grumman & Pigeon)	AA-5, AA-5A, AA-5B, AG-5B	A16EA	FAR 23					1/20/2005
28	Interceptor	See <i>Prop-Vets</i>							
29	Jodel	D-140-B	A3IN	CAR 10					
		DR-1050	A4IN	CAR 10					
		D-1190	A10IN	CAR 10					
		150	A10IN	CAR 10					
30	Lake (Revo)	C-1, C-2, LA-4, LA-4A, LA-4P, LA-4-200, 250	LA13	CAR3					1/10/2006
31	Lancair (Columbia)	LC40-50RG, LC42-50RG	A00603SE	FAR 23					10/17/2005

FAA Approved Model List (AML) SA01157LA

For Installation of Electronics International, Inc. Primary Fuel Flow/Pressure Instruments

Issue Date: June 1, 2001

Item	Aircraft Make	Aircraft Model	Original Type Certificate Number	Certification Basis For Alteration	Flight Manual Supplement		Installation Number	Instructions Revision	AML Amended Date
					Number	Revision			
32	Marchetti	S205-18F, S205-18R S205-20F, S205-20R S205-22R S208, S208A F250, F260B, F260C, F260D, F260E, F260F SIAI	A9EU	FAR 21	AFM2112	B 1/07/2003 or Later FAA Approved Revision	II S0506931	B 7/09/2004 or Later FAA Approved Revision	1/20/2005
			A9EU	FAR 21	"	"	"	"	"
			A9EU	FAR 21	"	"	"	"	"
			A9EU	FAR 21	"	"	"	"	"
			A10EU	CAR 3	"	"	"	"	"
33	Maule	M-4, M-4C, M-4-210, M-4-210C, M-4-210S, M-4-210T M-5-180C, M-5-200, M-5-210C, M-5-210TC, M-5-235C M-6-180, M-6-235 MXT-7-160, M-7-180, MX-7-180A, MX-7-180B, MX-7-180C MXT-7-180, MXT-7-180A M-7-235, M-7-235A, M-7-235B, M-7-235C, MT-7-235 M-8-235 BO-209-150 FV & RV BO-209-150 FV & RV BO-209-150 FF	3A23	CAR 3	"	"	"	"	"
			3A23	CAR 3	"	"	"	"	"
			3A23	CAR 3	"	"	"	"	"
			3A23	CAR 3	"	"	"	"	"
			3A23	CAR 3	"	"	"	"	"
			3A23	CAR 3	"	"	"	"	"
			3A23	CAR 3	"	"	"	"	"
			A27EU	FAR 21	"	"	"	"	"
			A27EU	FAR 21	"	"	"	"	"
			A27EU	FAR 21	"	"	"	"	"
34	Messerschmitt	See <i>Prop-142</i> M-18C, M-18C55, M-18L, M-18LA M20, M20A thru M20G, M20I, M20K, M20M, M20R, M20S M22	A-803	CAR 3	"	"	"	"	"
			2A3	CAR 3	"	"	"	"	"
			A6SW	CAR 3	"	"	"	"	"
			A31EU	FAR 21	"	"	"	"	"
35	Meyers	P-166, P-166B, P-166C P-136-L, P-136-L1, P-136-L2 PC-6, PC-6-HI, PC-6-H2 PC-6/350, PC-6/350-HI, PC-6/350-H2	7A4	CAR 10	"	"	"	"	"
			A-813	CAR 10	"	"	"	"	"
			7A15	CAR 10	"	"	"	"	"
36	Mooney Aircraft	PA-24-250, PA-24-260, PA-24-400 PA-28-180 PA-28-180, PA-28R-200 PA-28R-201, PA-28-201T, PA-28R-201T, PA-28-235 PA-28-235, PA-28R-201, PA-28R-201T, PA-28S-180 PA-30 PA-31, PA-31-300, PA-31-325, PA-31-350 PA-31P, PA-31P-350 PA-32-260, PA-32-300, PA-32R-300, PA-32RT-300 PA-32S-300, PA-32-301, PA-32-301T, PA-32R-301 PA-32R-301T, PA-34-200, PA-34-200T, PA-34-220T PA-36-285, PA-36-300, PA-36-375	7A15	CAR 3, 10	"	"	"	"	"
			1A15	CAR 3	"	"	"	"	"
			2A8	CAR 3	"	"	"	"	"
			2A13	CAR 3	"	"	"	"	"
			2A13	CAR 3	"	"	"	"	"
			2A13	CAR 3	"	"	"	"	"
			2A13	CAR 3	"	"	"	"	"
			A1EA	CAR 3	"	"	"	"	"
			A20S0	CAR 3	"	"	"	"	"
			A8EA	CAR 3	"	"	"	"	"
37	Navion	P-48, P-68B, P-68C, P-68C-TC P-166, P-166B, P-166C P-136-L, P-136-L1, P-136-L2 PC-6, PC-6-HI, PC-6-H2 PC-6/350, PC-6/350-HI, PC-6/350-H2	A31EU	FAR 21	"	"	"	"	"
			7A4	CAR 10	"	"	"	"	"
			A-813	CAR 10	"	"	"	"	"
38	Partenavia	P-166, P-166B, P-166C P-136-L, P-136-L1, P-136-L2 PC-6, PC-6-HI, PC-6-H2 PC-6/350, PC-6/350-HI, PC-6/350-H2	A-813	CAR 10	"	"	"	"	"
			7A15	CAR 10	"	"	"	"	"
			7A15	CAR 10	"	"	"	"	"
39	Piaggio	P-166, P-166B, P-166C P-136-L, P-136-L1, P-136-L2 PC-6, PC-6-HI, PC-6-H2 PC-6/350, PC-6/350-HI, PC-6/350-H2	A-813	CAR 10	"	"	"	"	"
			7A15	CAR 10	"	"	"	"	"
			7A15	CAR 10	"	"	"	"	"
40	Pilatus	P-166, P-166B, P-166C P-136-L, P-136-L1, P-136-L2 PC-6, PC-6-HI, PC-6-H2 PC-6/350, PC-6/350-HI, PC-6/350-H2	A-813	CAR 10	"	"	"	"	"
			7A15	CAR 10	"	"	"	"	"
			7A15	CAR 10	"	"	"	"	"
41	Piper Aircraft Co.	PA-24-250, PA-24-260, PA-24-400 PA-28-180 PA-28-180, PA-28R-200 PA-28R-201, PA-28-201T, PA-28R-201T, PA-28-235 PA-28-235, PA-28R-201, PA-28R-201T, PA-28S-180 PA-30 PA-31, PA-31-300, PA-31-325, PA-31-350 PA-31P, PA-31P-350 PA-32-260, PA-32-300, PA-32R-300, PA-32RT-300 PA-32S-300, PA-32-301, PA-32-301T, PA-32R-301 PA-32R-301T, PA-34-200, PA-34-200T, PA-34-220T PA-36-285, PA-36-300, PA-36-375	1A15	CAR 3	"	"	"	"	"
			2A8	CAR 3	"	"	"	"	"
			2A13	CAR 3	"	"	"	"	"
			2A13	CAR 3	"	"	"	"	"
			2A13	CAR 3	"	"	"	"	"
			2A13	CAR 3	"	"	"	"	"
			A1EA	CAR 3	"	"	"	"	"
			A20S0	CAR 3	"	"	"	"	"
			A8EA	CAR 3	"	"	"	"	"
			A3SO	CAR 3	"	"	"	"	"
42	Piper Aircraft Co.	PA-24-250, PA-24-260, PA-24-400 PA-28-180 PA-28-180, PA-28R-200 PA-28R-201, PA-28-201T, PA-28R-201T, PA-28-235 PA-28-235, PA-28R-201, PA-28R-201T, PA-28S-180 PA-30 PA-31, PA-31-300, PA-31-325, PA-31-350 PA-31P, PA-31P-350 PA-32-260, PA-32-300, PA-32R-300, PA-32RT-300 PA-32S-300, PA-32-301, PA-32-301T, PA-32R-301 PA-32R-301T, PA-34-200, PA-34-200T, PA-34-220T PA-36-285, PA-36-300, PA-36-375	A7SO	FAR 23	"	"	"	"	"
			A9SO	FAR 23	"	"	"	"	"
			A9SO	FAR 23	"	"	"	"	"

FAA Approved Model List (AML) SA01157LA

For Installation of

Electronics International, Inc. Primary Fuel Flow/Pressure Instruments

Issue Date: June 1, 2001

Item	Aircraft Make	Aircraft Model	Original Type Certificate Number	Certification Basis For Alteration	Flight Manual Supplement Number	Revision	Installation Instructions Number	Revision	AML Amended Date
42	Pitts	PA-39, PA-40	A1EA	CAR 3	AFM2112	B 1/07/2005 or Later FAA Approved Revision	II S050931	B 7/09/2004 or Later FAA Approved Revision	
43	Prop-Jets (Interceptor) (Aero Commander & Meyers)	PA-44-180, PA-44-180T PA-46-310P, PA-46-350P See Aviat	A19SO A25SO	FAR 23 FAR 23					
44	Revo	200, 200A, 200B, 200D, 403	3A18	CAR 3					
45	Rockwell	See Lake							1/10/2006
46	Sky International	See Commander Aircraft							1/20/2005
47	Socata Group (Aerospatiale)	TB 9, TB 10, TB 20, TB 21 TB 200 GA-7	A51EU A51EU A17SO	CAR 3 CAR 3 FAR 23					
48	Stinson	See Univair							
49	Swift	See Globe							10/17/2005
50	Thompson (Navion, North American)	Navion A, B, D, E, F, G, H L-17A, L-17B, L-17C	A-782 A-782	CAR 3 CAR 3					1/20/2005
51	Tiger Aircraft	See Gulfstream American See American Champion							
52	Trytek	108, 108-1, 108-2, 108-3, 108-5 V-77	A-767 A-774	CAR 3 CAR 4A					
53	Univair Aircraft (Alon, Erco, Forney, Mooney) (Stinson)	L-5, L-5B, L-5C, L-5D, L-5E, L-5F-1 See Angusair Inc.	A-764	CAR 4A					
54	Varga	PZL M20 C3	A68EU	FAR 21					
55	WSK- "PZL-Mielec" OBR								
---	End of List								

Amended Date: 1/20/2005; 10/17/2005; 1/10/2006

FAA Approved:



Acting Manager, Seattle Aircraft
Certification Office